NATIONAL ADVISORY COMMITTEE FOR AERONAUTICS

TECHNICAL NOTE 2479

TABLES OF EXACT LAMINAR-BOUNDARY-LAYER SOLUTIONS WHEN THE WALL IS POROUS AND FLUID PROPERTIES

ARE VARIABLE

By W. Byron Brown and Patrick L. Donoughe

Lewis Flight Propulsion Laboratory Cleveland, Ohio

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TABLES OF EXACT LAMINAR-BOUNDARY-LAYER SOLUTIONS

WHEN THE WALL IS POROUS AND FLUID PROPERTIES

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SUMMARY

The three partial differential equations of the laminar boundary layer for two-dimensional steady-state compressible flow have been transformed into two ordinary differential equations by the method of Pohlhausen, Falkner, and Skan. The ordinary equations include parameters for expressing the simultaneous effects of pressure gradient in the main-stream flow through a porous wall and property changes in the fluid due to large temperature differences between the wall and the free stream.

A total of 58 cases have been solved numerically by the method of Picard. The Euler number (nondimensional pressure-gradient parameter) ranges in value from 1 (stagnation-point value) to the negative values found at the laminar separation points. Three rates of flow through the porous wall were considered (including the impermeable case where the flow rate is 0). Five temperature ratios (stream temperature divided by wall temperature) were used: the uncooled and unheated case (temperature ratio of 1), two cooled cases (temperature ratios of 2 and 4), and (for the impermeable wall only) two heated cases (temperature ratios of 1/2 and 1/4). Velocity, weight-flow, and temperature distributions are tabulated as are the dimensionless stream function of Falkner and Skan and its derivatives and the dimensionless temperature function of Pohlhausen and its derivatives.

For each case, displacement, momentum, and convection thicknesses, as well as Nusselt number and coefficient of friction at the wall, were computed.

INTRODUCTION

A method of solving the laminar boundary-layer equations in which the fluid properties change with the temperature, the pressure varies along the main stream, and the cooling air flows through a porous wall is given in reference 1. Only temperature ratios (ratio of stream to wall temperature) greater than 1 (cooling) were considered therein. Since that time, additional solutions have been obtained for temperature ratios less than 1 (heating) for an impermeable wall.

Results of an investigation at the NACA Lewis laboratory are tabulated herein from solutions of different combinations of temperature ratios for heating and cooling, pressure gradients in the direction of the main flow, and coolant flows through the porous wall. These tables include velocity, weight-flow (product of density times velocity), and temperature distributions as well as the dimensionless stream and temperature functions and their derivatives. In addition, dimensionless forms of displacement, momentum, and convection boundary-layer thicknesses, Nusselt numbers, and wall friction coefficients are given for each case considered.

The numerical tables which give the distributions of the velocity and temperature functions are the work of Mrs. Helen C. Desmon and her associates.

SYMBOLS

The following symbols are used in this report:

C constant of proportionality
$$\frac{\tau_W}{c_{f,W}} = \frac{\frac{\tau_W}{\rho_W U_\infty^2}}{\frac{2}{2}}$$

$$c_{f,\infty} = \frac{\frac{\tau_W}{\rho_\infty U_\infty^2}}{\frac{\rho_\infty U_\infty^2}{2}}$$

specific heat at constant pressure

Eu Euler number,
$$\frac{-x}{\rho_{\infty}U_{\infty}^2}$$
; $U_{\infty} = Cx^{Eu}$

f dimensionless stream function.

f',f",f" first, second, and third derivatives of f with respect to η

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k thermal conductivity

Nu Nusselt number, $\frac{Hx}{k_w}$

 $P \qquad \frac{T}{T_{w}} = 1 + \theta \left(\frac{T_{\infty}}{T_{w}} - 1 \right) = P$

p pressure

Pr Prandtl number, $\frac{c_{pw} \mu_w}{k_w}$

Re Reynolds number, $\frac{\omega^p w^2}{\mu_w}$

T fluid temperature

Two refers to wall temperature and coolant upon emergence from porous wall

Um fluid velocity at edge of boundary layer

u fluid velocity in boundary layer in x-direction parallel to wall

v fluid velocity in boundary layer in y-direction normal to wall

x distance along surface

y distance normal to surface

 α exponent of temperature for specific heat, $c_p \otimes T^{\alpha}$

 β pressure-gradient parameter, $\frac{2Eu}{Eu+1}$

δ* displacement boundary-layer thickness

 δ_c convection boundary-layer thickness

δi momentum boundary-layer thickness

 ϵ exponent of temperature for thermal conductivity, $k \otimes T^{\epsilon}$

η dimensionless boundary-layer coordinate, $y\sqrt{\frac{\rho_w}{\mu_w}}\frac{V_w}{x}$

heta temperature-difference ratio, $\frac{T-T_W}{T_{\infty}-T_W}$

(4)

 θ ', θ " first and second derivatives of θ with respect to η

μ absolute viscosity of fluid

ρ density of fluid

τ, shear stress at wall

♥ stream function

 ω exponent of temperature for viscosity, μ ∞ T^{ω}

Subscripts:

w wall

main stream

ANALYSIS

The equations of the laminar boundary layer for steady-state flow of a viscous fluid with heat transfer may be obtained from reference l as

Momentum equation:

$$\rho u \frac{\partial u}{\partial x} + \rho v \frac{\partial u}{\partial y} = \frac{\partial}{\partial y} \left(\mu \frac{\partial u}{\partial y} \right) - \frac{\partial p}{\partial x}$$
 (1)

Continuity equation:

$$\frac{\partial(\rho u)}{\partial x} + \frac{\partial(\rho v)}{\partial y} = 0 \tag{2}$$

Energy equation:

$$c_{p} \left(\rho u \frac{\partial T}{\partial x} + \rho v \frac{\partial T}{\partial y} \right) = \frac{\partial}{\partial y} \left(k \frac{\partial T}{\partial y} \right) + \mu \left(\frac{\partial u}{\partial y} \right)^{2} + u \frac{\partial p}{\partial x}$$
 (3)

The boundary conditions are: when y = 0,

$$u = 0$$
 $v = v_w$ $T = T_w$

and when $y \rightarrow \infty$,

$$u \to U_{\infty}$$
 $\frac{\partial u}{\partial y} \to 0$ $T \to T_{\infty}$ $\frac{\partial T}{\partial y} \to 0$

Assumptions

In order to simplify the analysis, the following assumptions are made:

- (1) The Mach number is small.
- (2) The Euler number is constant.
- (3) The wall temperature is constant.
- (4) The fluid property variations are expressible as some power of the absolute temperature:

$$\mu \otimes T^{(k)} \qquad k \otimes T^{\epsilon} \qquad c_p \otimes T^{\alpha} \qquad \rho \otimes T^{-1}$$
 (5)

Transformation to Ordinary Differential Equations

The transformation from partial to total differential equations is accomplished by the change in variables

$$\eta = y \sqrt{\frac{\rho_{W} U_{\infty}}{\mu_{X} x}}$$

$$\theta = \frac{T - T_{W}}{T_{\infty} - T_{W}}$$
(6)

and

$$\mathbf{f} = \frac{\rho_{\mathbf{W}} \, \, \boldsymbol{\Psi}}{\sqrt{\mu_{\mathbf{W}} \mathbf{X} \, \, \boldsymbol{U}_{\infty} \rho_{\mathbf{W}}}}$$

where η is the dimensionless independent variable introduced by Blasius, and f and θ are the dimensionless dependent variables representing the stream function and temperature, respectively.

Substitution of η , f, and θ in the partial differential equations and use of the simplifying assumptions yield (reference 1) the energy equation

$$-\theta'' = \frac{\text{Eu+l}}{2} \operatorname{Pr}_{W} \operatorname{P}^{\alpha-\epsilon} \theta' f + \epsilon \left(\frac{T_{\infty}}{T_{W}} - 1 \right) \operatorname{P}^{-1} \theta'^{2}$$
 (7)

and the momentum equation

and when $\eta \rightarrow \infty$,

$$f''' = \operatorname{Eu} P^{-\omega} f'^{2} - \frac{\operatorname{Eu}+1}{2} P^{-\omega} ff'' - \operatorname{Eu} \frac{T_{w}}{T_{\infty}} P^{-\omega-1} - \frac{\operatorname{Eu}+1}{2} \left(\frac{T_{\infty}}{T_{w}} - 1\right) P^{-\omega-1} ff'\theta' - \left(\frac{T_{\infty}}{T_{w}} - 1\right) P^{-1}f'\theta'' - \frac{\operatorname{Eu}+1}{2} \left(\frac{T_{\infty}}{T_{w}} - 1\right) P^{$$

$$(\omega+2)\left(\frac{\mathbb{T}_{\mathbf{w}}}{\mathbb{T}_{\mathbf{w}}}-1\right)P^{-1}f''\theta'-\omega\left(\frac{\mathbb{T}_{\mathbf{w}}}{\mathbb{T}_{\mathbf{w}}}-1\right)^{2}P^{-2}f'\theta'^{2}$$
(8)

The boundary conditions are: when $\eta = 0$,

$$f' = 0 f = f_{W} \theta = 0$$

$$\theta \to 1 \theta' \to 0 f' \to \frac{T_{W}}{T_{\infty}} f'' \to 0$$
(9)

CALCULATION OF TABLES

In the solution of equations (7) and (8), air in the range of 600° to 2400° F was chosen as the fluid for the main stream and the coolant forced through the porous wall was assumed to be air. Thus \Pr_{W} was taken at 0.7, the exponent ω in the viscosity-temperature relation was 0.7, the exponent ε in the thermal-conductivity temperature relation was 0.85, and the exponent α in the specific heat-temperature relation was 0.19.

In the first six cases (table I(1)), the velocity distributions had already been calculated in reference 2. In these cases, the wall was impermeable ($f_W = 0$) and the fluid properties were constant $(T_{\infty}/T_W = 1)$. The energy equation (1) thus reduced to the simple form

$$-\theta'' = \frac{\text{Eu+l}}{2} \text{Pr}_{\mathbf{W}} \theta' \mathbf{f} \tag{10}$$

This is a linear equation of the first order and hence is readily solved after f is found by a numerical integration of the velocity distributions. The solution given in reference 3 is

$$\theta = \frac{\int_{0}^{\eta} e^{-\frac{\operatorname{Eu}+1}{2} \operatorname{Pr}_{W} \int_{0}^{\eta} f \, d\eta}}{\int_{0}^{\infty} e^{-\frac{\operatorname{Eu}+1}{2} \operatorname{Pr}_{W} \int_{0}^{\eta} f \, d\eta}}$$
(11)

The Blasius and Pohlhausen distributions are given in table I(2).

Then θ and u/U_{∞} were tabulated as functions of η . The value of Nu//Re was given by the reciprocal of the denominator of equation (11); that is

$$\frac{Nu}{\sqrt{Re}} = \frac{1}{\int_{0}^{\infty} e^{-\frac{Eu+1}{2} \operatorname{Pr}_{W} \int_{0}^{\eta} f \, d\eta}} = \theta_{W}'$$
 (12)

The value of f_w'' was taken from reference 2.

The three thicknesses were computed from the equations derived in reference 1:

$$\frac{\delta^*_{\overline{X}}}{\sqrt{Re}} = \int_0^\infty \left(1 - \frac{T}{T_W} f'\right) d\eta$$

$$\frac{\delta_1}{\overline{X}} \sqrt{Re} = \frac{T_{\infty}}{T_W} \int_0^\infty f'(1-Pf') d\eta$$

$$\frac{\delta_C}{\overline{X}} \sqrt{Re} = \frac{T_{\infty}}{T_W} \int_0^\infty f'(1-\theta) d\eta$$
(13)

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The remaining parts of table I contain the values of f, f', f', and f'' and θ , θ ', and θ ". These values were obtained by solving equations (7) and (8) numerically by the method of Picard, as explained in reference 1. The thicknesses were computed by equations (13) as before.

When the temperature ratio was 1, u/U_{∞} was given by the f'table, as follows from equation (6) with P=1. When $T_{\infty}/T_{w}\neq 1$,

$$\frac{\mathbf{u}}{\mathbf{U}_{\infty}} = \mathbf{f}' \left[1 + \theta \left(\frac{\mathbf{T}_{\infty}}{\mathbf{T}_{w}} - 1 \right) \right] = \mathbf{f}' \mathbf{P} \tag{14}$$

and $\rho u/\rho_{\infty}U_{\infty}$ was given by f' T_{∞}/T_{W} obtained by multiplying equation (14) by ρ/ρ_{∞} or T_{∞}/T .

From reference 1 also,

$$\frac{\text{Nu}}{\sqrt{\text{Re}}} = \theta_{\text{W}}^{1}$$

A summary of the principal parameters is given in table II, which serves as an index to table I. Here all the thicknesses are grouped together with the values of Nu/ \sqrt{Re} and of $f_W^{"}$. If C_f is defined by the equation

$$\tau_{\rm W} = \frac{1}{2} \, {\rm C_{f,W}} \, \rho_{\rm W} {\rm U_{\infty}}^2$$

then

$$f_{W}^{"} = \frac{C_{f,W}}{2} \sqrt{Re}$$

Conversely, if $C_{\mathbf{f}}$ is defined by the relation

$$\tau_{W} = \frac{1}{2} C_{f,\infty} \rho_{\infty} U_{\infty}^{2}$$

then

$$f_{\mathbf{w}}^{"} = \frac{c_{f,\infty}}{2} \frac{T_{\mathbf{w}}}{T_{\infty}} \sqrt{Re}$$

DISCUSSION

For the case of the heated wall $(T_{\infty}/T_{\rm W}=1/2,\ 1/4)$, the boundary-layer velocity was found to be higher than the stream velocity for Euler numbers 0.5 and 1 with no coolant flow. This apparent anomaly is probably due to the high pressure gradient imposed on the flow in conjunction with the large amount of heating at the wall. The velocity distributions for heating and cooling are shown in figure 1 for the impermeable wall and an Euler number of 1. The calculation for the momentum thicknesses yielded negative values for temperature ratio of 1/4 with Eu = 0.5 and 1.0 (table II).

The values of the Euler number at the separation point are plotted against the temperature ratio for the impermeable wall in figure 2. This curve is obtained by setting $f_W^*=0$ for each temperature ratio. The value for temperature ratio of 1 was obtained by Hartree (reference 2). Increasing the temperature ratio from 1 to 4 permits a 50 percent greater adverse pressure gradient, whereas reducing it from 1 to 1/4 decreases the permissible adverse pressure gradient by about 60 percent.

CONCLUDING REMARKS

Calculations of 58 velocity and temperature distributions were made for air with a low Mach number by equations that include the simultaneous effects of pressure gradients in the main stream, flow through a porous wall, and large temperature variations through the boundary layer.

The pressure gradients vary from stagnation point values (Euler number of 1) to the values occurring at the laminar separation point. Three rates of flow through the porous wall, represented by 0, -0.5, and -1, have been used; three values of the ratio of stream to wall temperature have been used throughout, 1, 2, and 4. For the impermeable wall, two additional ratios were used, 1/2 and 1/4.

For each case, a complete tabulation throughout the boundary layer was made of velocity, weight-flow, and temperature distributions, as well as of the nondimensional stream and temperature functions and their derivatives.

Each case determines a value for the Nusselt number, the wall friction coefficient, and the displacement, momentum, and convection thicknesses.

When the wall is much hotter than the stream and large favorable pressure gradients exist, boundary-layer velocities may exceed the free-stream values by amounts ranging as high as 20 percent (near the stagnation point when the wall temperature is four times the stream temperature). For some of the heated-wall profiles, the momentum thicknesses are negative; that is, the boundary layer contains more energy than a corresponding section of the free stream.

Lewis Flight Propulsion Laboratory, National Advisory Committee for Aeronautics, Cleveland, Ohio, May 31, 1951.

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					011000 1112			
			(1) f _W =	= 0; T _{∞/}	$T_W = 1$		NACA	-
Nu √Re	= 0.2	231	Nu √Re	= 0.	221	Nu √Re	= 0.	198
f"	= .0	87	f"	= .(058	f"	= 0	
<u>δ*√</u>	Re = 2.7	62	<u>δ*√1</u>	Re = 2.9	972	<u>δ*√</u>	Re = 3.	498
$\frac{\delta_1 \sqrt{x}}{x}$	Re8	38	$\frac{\delta_{1}\sqrt{1}}{x}$	Re = .8	353	$\frac{\delta_1 \sqrt{x}}{x}$	Re	868
$\frac{\delta_{c}\sqrt{x}}{x}$	Re = .7	20	δ _c √I	<u>₹e</u> = .6	393	δ _c √.	Re = .	626
Eu=-0.	0826 β=	-0.18	Eu=-0.	0868 в	=-0. 19	Eu=-0.09	04 β=-	0.1988
η	θ	u/U_{∞}^{-1}	η	θ	u/U_{∞}^{-1}	η	. 0	u/V _∞ 1
0 .295 .591 .886 1.181 1.476 2.362 2.658 2.953 3.248 3.543 3.543 3.539 4.129 4.725 5.315 5.611 5.201 6.492 7.087 7.382 7.678 7.978 8.563 8.859 9.154 9.745 10.040	0 .068 .136 .204 .272 .340 .406 .471 .534 .595 .652 .706 .756 .801 .840 .874 .903 .927 .946 .962 .973 .981 .999 .999 .999 .999 .000 1.000 1.000 1.000 1.000 1.000 1.000 1.000	0 .029 .066 .109 .160 .217 .279 .346 .417 .490 .562 .633 .700 .760 .815 .861 .899 .929 .952 .968 .980 .988 .998 .998 .998 .999 .999 .99	0 .296 .592 .888 1.184 1.480 1.776 2.072 2.368 2.664 2.960 3.256 3.552 3.848 4.144 4.440 4.736 5.032 5.328 5.624 5.920 6.216 6.512 6.808 7.104 7.400 7.696 7.992 8.288 8.584 8.880 9.176 9.472 9.768 10.064 10.360	0 .066 .131 .196 .262 .326 .391 .454 .515 .575 .632 .859 .890 .916 .937 .954 .967 .998 .999 .999 .999 .999 .999 .999 .99	0 .021 .050 .086 .129 .179 .236 .299 .366 .437 .510 .581 .716 .775 .871 .907 .935 .956 .972 .989 .994 .996 .998 .998 .999 1.000 1.000 1.000	0 .297 .593 .890 1.186 1.483 1.779 2.076 2.372 2.669 2.966 3.262 3.559 3.855 4.152 4.448 4.745 5.042 5.338 5.635 5.931 6.228 6.821 7.117 7.414 7.711 8.007 8.304 8.897 9.193 9.490 9.786 10.083 10.380 10.676 10.973	.118 .177 .236 .295 .354 .469 .526 .580 .633 .684 .731 .775 .815 .850	.016 .036 .064 .099 .142 .193 .250 .313 .380

¹Hartree

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(1) $f_W = 0$; $T_{\infty}/T_W = 1$ - Concluded

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Ni √Re	$\frac{1}{2} = 0.1$	267	Nu √Re	= 0.2	253	Nu √R∈	= 0	.243
f"	= .:	2 2 0	f"			f _W " = .130		
<u>δ*</u> _Δ	$\frac{\sqrt{\text{Re}}}{\kappa} = .0$	092	<u>δ*√</u> x	Re = 2.	336	<u>δ*,</u>	$\frac{\sqrt{\text{Re}}}{c} = 2$.510
$\frac{\delta_1 \sqrt{Re}}{x} = .746$			$\frac{\delta_1\sqrt{x}}{x}$	Re _	788	$\frac{\delta_{1\sqrt{x}}}{x}$	/ <u>Re</u>	812
δcu	/Re = .8	301	$\frac{\delta_{c}\sqrt{x}}{x}$	Re =	7 7 3	$\frac{\delta_{c}\sqrt{x}}{x}$	Re =	752
Eu=-0.	0476 β	=-0.10	Eu=-0.	0654	β=-0.14	Eu=-0.	0741 · p	=-0.16
η	. , θ	u/U _∞ 1	η	θ	u/U∞ ¹	η	θ`	u/U∞ ¹
0 .290 .580 .869 1.159 1.449 1.739 2.319 2.608 2.188 3.478 3.768 4.057 4.637 4.637 5.217 5.5796 6.666 6.956 7.245 7.535 7.825 8.105 8.405 8.984 9.274	0 .077 .155 .232 .308 .383 .456 .527 .594 .658 .716 .814 .854 .888 .916 .938 .956 .969 .979 .986 .999 .998 .999 .999 .999 .999 .99	0 .066 .136 .209 .285 .363 .442 .519 .595 .666 .731 .790 .840 .882 .915 .941 .991 .994 .997 .998 .999 .999 1.000 1.000	0 .293 .585 .878 1.170	.746 .794 .836 .872 .902 .927 .947	0 .051 .107 .168 .235 .305 .378 .453 .528 .602 .671 .735 .793 .842 .884 .917 .942 .962 .975 .984 .991 .995 .997 .998 .999 1.000 1.000 1.000	0 .294 .588 .882 1.176 1.470 1.764 2.058 2.646 2.352 2.646 2.939 3.233 3.527 3.821 4.115 4.409 4.703 4.997 5.291 5.879 6.467 6.761 7.055 7.349 7.643 7.936 8.230 8.524 8.818 9.406 9.700 9.994	.143 .214 .285 .356 .425 .557 .619 .677 .730 .779 .822 .860 .892 .918 .939 .956	.089 .143 .202 .267

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(2)
$$T_{\infty}/T_{W} = 1$$
; Eu = 0; $f_{W} = 0$ 1
 $\frac{\delta^{*}\sqrt{Re}}{x} = 1.721$; $\frac{\delta_{1}\sqrt{Re}}{x} = 0.662$; $\frac{\delta_{c}\sqrt{Re}}{x} = 0.834$

f" f' f" θ" η f θ 01 0 0.332 0.293 .4 .027 .332 -.004 .292 .133 .117 -.003 .106 .264 -.011 .8 .328 -.017 .234 .290 -.038 .238 1.2 .394 .316 .348 -.024 .283 -.040 1.6 .420 .517 .297 -.062 .459 .270 2.0 .650 .630 -.087 .267 .564 .251 -.057 2.4 .922 .729 .228 -.105 .659 .225 -.073 2.8 1.231 .812 .184 -.113 .194 -.083 .743 1.569 .876 .139 -.109 .814 .159 -.087 3.6 .098 -.095 1.930 .924 .870 .125 -.084 4.0 2.306 .956 -.074 .064 .914 .093 -.075 2.692 -.052 .976 .039 .945 .065 -.062 4.8 3.085 .988 .022 -.034 .967 .044-.047 5.2 3.481 .994 .011 -.020 .981 .028 -.034 5.6 3.880 .998 .006 -.010 .989 .016 - .022 6.0 4.280 .999 .002 -.005 -.014 .994 .009 4.679 1,000 .997 6.4 .001 -.002 .005 -.008 6.8 1.000 5.079 -.001 .000 .999 .002 -.004 7.2 5.479 1.000 .000 .000 .999 .001 -.002 1.000 7.6 5.879 .000 .000 1.000 •000 -.001 8.0 6.279 1.000 .000 .000 1.000 .000 .000 6.679 1.000 .000 .000 8.8 7.079 1.000 .000 .000 9.2 7.479 1.000 .000 .000

^{1&}lt;sub>Blasius</sub>

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(3)
$$T_{\infty}/T_{W} = 1$$
; Eu = 0.5; $f_{W} = 0$

$$\frac{\delta^* \sqrt{Re}}{x} = 0.855; \frac{\delta_1 \sqrt{Re}}{x} = 0.374; \frac{\delta_c \sqrt{Re}}{x} = 0.792$$

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			•		,			
	η	f	f'	f"	f"	θ	91	θ"
111122222333333444445555566666	246802468024680246802468	0 .017 .044 .368 .56827 1.3762 1.3762 1.3752 1.3752 1.3752 1.3753 1.345 1.35 1.35 1.35 1.35 1.35 1.35 1.35 1.3	0 .170 .320 .451 .563 .658 .737 .801 .852 .923 .946 .963 .976 .984 .990 .994 .998 .999 .999 .000 1.000 1.000 1.000	0.900 .800 .702 .607 .517 .433 .356 .287 .227 .176 .134 .099 .072 .051 .035 .024 .016 .010 .006 .004 .002 .001 .000	-0.500496484464437403364322278234192154120091067048034023015009006004002001000000	0 .083 .166 .249 .330 .409 .485 .557 .624 .686 .742 .791 .833 .870 .900 .944 .959 .970 .986 .997 .998 .999 .999 .999 .999 .999 .999	0.416 .416 .414 .410 .401 .389 .323 .294 .262 .229 .196 .165 .136 .109 .086 .067 .051 .038 .027 .019 .004 .009 .004 .009 .000	0004014031052075099121140154163165161152139123107089073058045034025018009006004002002001001 .000 .000 .000 .000

(4)
$$T_{\infty}/T_{W} = 1$$
; Eu = 1; $f_{W} = 0$

$$\frac{\delta^{4}\sqrt{Re}}{x} = 0.648; \frac{\delta_{1}\sqrt{Re}}{x} = 0.290; \frac{\delta_{c}\sqrt{Re}}{x} = 0.708$$

η	f	η	
1234567890123456789012345678901234	0 .00311 .00531	.2 .3 .4 .5 .6 .7	

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(5)
$$T_{\infty}/T_{W} = 2$$
; Eu = -0.1178; $f_{W} = 0$

$$\frac{\delta^{*}\sqrt{Re}}{x} = 4.582$$
; $\frac{\delta_{1}\sqrt{Re}}{x} = 1.664$; $\frac{\delta_{C}\sqrt{Re}}{x} = 1.076$

η	f	f'	f"	f'''	θ	θ'	θ"	u/U∞	ρυ ρ _ω U _∞
0 2468024682604826048260482604826048260482	0 .000 .001 .002 .004 .008 .014 .021 .030 .042 .056 .073 .092 .114 .139 .197 .268 .350 .445 .552 .670 .800 .940 1.248 1.414 1.587 1.766 1.949 2.137 2.328 2.717 2.914 3.112 3.510 3.710 3.909 4.709 4.709 4.509 4.709 5.109 5.309 5.	0 .001 .004 .009 .016 .023 .032 .042 .053 .064 .077 .090 .117 .131 .161 .191 .222 .252 .310 .337 .362 .385 .406 .424 .440 .453 .464 .474 .481 .486 .491 .494 .496 .498 .499 .500 .500 .500 .500 .500 .500 .500	0 .011 .020 .028 .035 .042 .047 .052 .056 .060 .063 .071 .072 .075 .076 .075 .076 .075 .070 .065 .020 .016 .012 .009 .007 .005 .000 .000 .000 .000 .000 .000	0.059 .050 .043 .038 .029 .026 .020 .018 .016 .014 .012 .010 .008 .005 .002 001 004 007 015 015 015 015 015 015 015 015 016 004 007 001 004 007 001 004 007 001 001 004 007 001 001 001 001 002 001 000 001 000 001 000 001 000 001 000 000 001 000 	0 .037 .073 .109 .143 .176 .209 .241 .2302 .3361 .390 .417 .548 .643 .760 .793 .824 .851 .876 .898 .916 .933 .946 .958 .997 .998 .999 .999 .999 .999 .999 .99	0.189 .183 .178 .173 .169 .165 .161 .157 .154 .150 .147 .144 .141 .138 .135 .129 .123 .116 .103 .095 .088 .080 .073 .065 .088 .051 .044 .037 .032 .026 .022 .017 .014 .011 .008 .006 .005 .004 .007 .001 .000 .000 .000	-0.030028025023021020019016016015015015015015016016017018019019019019019019019019019019019019019010008007005004003002001001000000	0 .001 .004 .010 .018 .027 .039 .052 .067 .084 .102 .143 .166 .190 .241 .296 .354 .414 .474 .535 .869 .835 .752 .796 .835 .752 .796 .835 .941 .957 .969 .998 .999 .998 .999 .998 .999 .998 .999 .998 .999 .990 .990 .990 .990 .900 .900 .900 .900 .900 .900 .900 .900 .900	0 .002 .008 .018 .031 .046 .064 .106 .129 .153 .206 .234 .262 .322 .382 .443 .504 .621 .675 .771 .812 .848 .880 .907 .929 .947 .947 .958 .999 .999 .999 .999 .999 .999 .999

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(6)
$$T_{\infty}/T_{W} = 2$$
; Eu = -0.09; $f_{W} = 0$

$$\frac{\delta^* \sqrt{\text{Re}}}{x} = 2.430; \frac{\delta_1 \sqrt{\text{Re}}}{x} = 1.501; \frac{\delta_c \sqrt{\text{Re}}}{x} = 1.408$$

				A				N. N.	ACA
η	f	f¹	f"	f"†	θ	θ'	θ"	u/U _∞	ρυ ρ ω υ _∞
0 123456.78902468260482611.22222233344485.664208642086421228642213344.42	0 .001 .003 .007 .019 .028 .037 .048 .060 .073 .138 .178 .2268 .375 .434 .496 .774 .929 1.264 1.442 1.625 2.392 2.787 3.186 3.585 3.185 5.585 5.985 6.385	0 01316 004615 00758 011479 0112363578864 011479	0.163 .157 .152 .146 .142 .138 .130 .127 .121 .115 .110 .106 .101 .097 .088 .080 .071 .098 .080 .071 .040 .033 .027 .009 .000 .000 .000	-0.066059053048044040037034029026024023022021021021021021021021021021021020019018016015011008004002001000	0 025 049 073 0123 1165 1203 1231 1350 1459 1231 1350 1459 1555 1666 1761 1835 1946 1999 1999 1000 1	0.252 .247 .242 .237 .233 .229 .225 .221 .217 .214 .210 .203 .197 .190 .184 .172 .165 .153 .141 .128 .1164 .092 .080 .070 .050 .034 .002 .001 .000 .000 .000	-0.05405104804504503903803703603503103000290028006003009006000000000	0 .016 .033 .050 .067 .084 .101 .118 .136 .154 .171 .207 .243 .280 .316 .352 .389 .425 .460 .495 .5626 .685 .739 .787 .828 .864 .920 .956 .977 .989 .995 .995 .999 .995 .995 .999 .995 .999 .995 .999 .995 .999 .995 .995 .999 .995 .990 .995 .995	0 .032 .063 .093 .1249 .177 .203 .229 .254 .278 .326 .371 .414 .455 .533 .569 .636 .637 .750 .798 .839 .874 .903 .993 .998 .999 1.000

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(7)
$$T_{\infty}/T_{W} = 2$$
; Eu = -0.05; $f_{W} = 0$

$$\frac{\delta^* \sqrt{Re}}{x} = 1.882; \frac{\delta_1 \sqrt{Re}}{x} = 1.383; \frac{\delta_c \sqrt{Re}}{x} = 1.478$$

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η	f	f'	ſ"	f"t	θ	θι	θ"	u/U∞	ρυ ρ _∞ U _∞
0 1234568024682604826864208664 1112222233344455567889011223344 11223334445556788911223344455567889011223344445556788901122334444555678890112233444455567889011223344445556788901122334444555678890112233444455567889011223344445556788901122334444555678890112233444455567889011223344445556788901122334444555678890112233444455567889011223344445556788901122334444555678890112233444455567889011223344445556788901122334444556788901122334444556788901122334444556788901122334444689011223344445567889001122334444689011223344446890112233444468901122334444689011223344446890112233444468901122334444689011223344448900000000000000000000000000000000	0 .0015 .010 .018 .0289 .0527 .0842 .1488 .2354 .4185 .4185 .4185	0 .024 .046 .067 .086 .124 .141 .157 .173 .188 .2142 .2268 .33245 .345 .425 .4458 .498 .498 .499 .500	0.243 .229 .216 .204 .184 .175 .167 .163 .147 .135 .125 .107 .099 .091 .078 .072 .060 .050 .041 .033 .026 .020 .000	-0.156138123110091083076071066054049045041038036034032030027024019016014011007004001	0 027 .054 .080 .105 .130 .226 .249 .233 .249 .293 .336 .377 .415 .560 .592 .706 .754 .797 .834 .865 .934 .962 .979 .989 .999	0.276 .269 .264 .258 .253 .248 .230 .226 .218 .210 .202 .194 .187 .179 .172 .164 .157 .142 .127 .113 .099 .086 .073 .062 .027 .017 .009 .005 .001 .000 .000	-0.065060057053051048046045042041040039038038037004001001001000000000	0 .048 .0726 .0193 .1469 .1439	0 .047 .092 .134 .173 .211 .247 .281 .314 .532 .484 .532 .576 .655 .690 .723 .753 .806 .850 .938 .956 .970 .987 .995 .999 .999

(8)
$$T_{\infty}/T_{W} = 2$$
; Eu = 0; $f_{W} = 0$
 $\frac{\delta^{*}\sqrt{Re}}{x} = 1.537$; $\frac{\delta_{1}\sqrt{Re}}{x} = 1.271$; $\frac{\delta_{C}\sqrt{Re}}{x} = 1.495$

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1		•						w.	ALL .
η	f	f†	f"	. f ^m	θ	θ 1	θ"	u∕U∞	ρυ ρωυ _∞
0 .123.45.66.78.90.24.68.26.04.48.26.64.20.8 11.22.22.23.34.48.25.66.4.20.8 10.12.8 11.22.8	0 .002 .006 .013 .023 .035 .049 .065 .083 .125 .173 .287 .351 .491 .567 .646 .728 .898 1.267 .1450 1.450 1.643 2.432 2.832 2.432 2.832 2.432 2.632 2.432 2.6	0 .058 .084 .132 .199 .2256 .235 .33570 .414 .436 .4458 .4499 .499 .500 .500	0.312 .289 .269 .250 .234 .183 .163 .146 .1318 .106 .096 .077 .049 .030 .011 .000 .000 .000 .000	-0.248218193172154139127116098091079070062056050046042038035029025020017014011008002000	0 .029 .057 .085 .112 .139 .165 .191 .216 .240 .264 .311 .356 .399 .440 .479 .517 .553 .588 .620 .681 .734 .781 .822 .857 .912 .949 .972 .986 .994 .998 1.000 1.000 1.000 1.000 1.000 1.000	0.294 .287 .281 .274 .269 .263 .258 .252 .247 .243 .220 .211 .203 .185 .168 .159 .1426 .110 .095 .000 .000 .000 .000	-0.074 -068 -060 -057 -055 -055 -055 -049 -048 -045 -044 -043 -043 -043 -043 -043 -043 -043	0 .031 .061 .091 .149 .177 .225 .235 .3385 .437 .520 .637 .734 .787 .835 .926 .946 .976 .998 .999 .999 .999 .999 .999 .999 .99	0 .060 .1168 .2162 .304 .3442 .3344 .5168 .663 .740 .7773 .829 .9356 .9356 .9356 .9999 .9999 .9999 .9999

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(9)
$$T_{\infty}/T_{W} = 2$$
; Eu = 0.5; $f_{W} = 0$

$$\frac{\delta^{*}\sqrt{Re}}{x} = 0.699$$
;
$$\frac{\delta i\sqrt{Re}}{x} = 0.899$$
;
$$\frac{\delta c\sqrt{Re}}{x} = 1.370$$

NACA ρu f" f^{III} O 1 01 u/U∞ PooUo θ f f' η 0.400 -0.136 0.679 -0.984 0 0 -.114 -.101 .23.7 .128 .514 .376 -.700 .078 .012 .118 .240 .417 .151 .394 .355 -.515 .208 .4 .046 .556 -.093 .339 .303 -.388 .220 .335 .277 .6 .094 .425 -.088 -.296 .316 .155 .235 .284 .8 .331 .745 -.230 .299 -.086 .501 .346 1.0 .226 .372 .183 -.084 .568 .809 .282 -.180 .404 .405 .142 1.2 .304 -.083 .860 .459 .265 .627 .430 .110 -.142 1.4 .387 .898 -.082 .678 .510 .249 -.112 .084 1.6 .475 .449 .723 .232 -.081 .928 -.089 .558 .064 .464 1.8 .566 -.080 .950 .216 .762 -.071 .603 .048 .475 2.0 .660 .200 .967 .645 -.078 .795 -.057 .036 .756 .484 2.2 -.076 -.044 .683 .185 .824 .979 .026 .490 2.4 .854 .170 -.074 -.036 .849 .988 .719 .752 .494 .018 .952 2.6 .156 -.071 .870 .994 .497 1.051 .011 -.028 2.8 .142 - .068 .888 -.022 1.151 .498 .006 .781 3.0 -.065 -.061 -.058 .903 .999 .128 .808 -.017 .499 .002 3.2 1.250 .916 .999 .116 -.013 .833 .500 .000 3.4 1.350 .927 .104 .855 1.450 3.6 -.050 .082 .946 .892 4.0 1.650 -.042 .961 .921 .064 1.850 4.4 -.035 .972 .049 .944 2.050 4.8 .980 .036 -.028 .960 2.250 -.022 .026 .986 .973 2.450 5.6 -.017 .982 .018 .991 6.0 2.650 -.012 .994 .988 .013 6.4 2.850 -.009 .996 .992 .009 3.050 6.8 -.006 .997 .006 3.250 .995 7.2 .004 -.004 .998 .997 3.450 7.6 -.003 .999 .998 .002 8.0 3.650 .999 .001 -.002 .999 3.850 8.4 -.001 1.000 .999 .001 8.8 4.050 .001 .999 -.001 1.000 4.250 9.2 -.001 1.000 1.000 .000 9.6 4.450 .000 .000 1.000 1.000 10.0 4.650 .000 .000 1.000 1.000 4.850 10.4

(10)
$$T_w/T_w = 2$$
; Eu = 1; $f_w = 0$
 $\frac{\delta^*/\overline{Re}}{x} = 0.515$; $\frac{\delta_1/\overline{Re}}{x} = 0.763$; $\frac{\delta_c/\overline{Re}}{x} = 1.215$

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η	f	f'	f"	f"	θ	01	θ"	ս/Մ_	$\rho u \over \rho_{\infty} U_{\infty}$
0 .12345678901246802468024680482604826048	0 .016 .034 .005 .015 .005 .115 .227 .159 .2315 .444 .742 .242 .242 .242 .242 .242 .242	0 .051 .257 .293 .368 .425 .445 .4497 .499 .500	0.899 .750 .629 .529 .446 .376 .318 .268 .226 .190 .160 .134 .112 .077 .051 .032 .018 .007	-1.647 -1.336 -1.098 910 760 637 537 455 386 239 205 150 111 082 060 044 032	0 .046 .091 .176 .2155 .293 .365 .399 .463 .523 .523 .625 .718 .756 .790 .847 .891 .925 .938 .948 .995 .997 .998 .999 .999 .999 .999 .999 .999	0.473 .455 .438 .424 .410 .396 .383 .371 .359 .347 .335 .323 .311 .288 .265 .243 .221 .200 .180 .161 .143 .126 .111 .096 .083 .072 .061 .043 .030 .020 .013 .000 .000 .000 .000	-0.190169154136130126124122120119118117115113110107102098092087081074068056050039029021015010006001001001000000	0 .1657 .3063 .418 .5166 .5950 .6662 .7173 .68357 .68357 .7601 .88577 .8910 .3555 .68357 .8910 .9956 .9956 .9999 .9999 .9999 .9999 .9999 .9999 .9999	0 .165 .302 .418 .515 .597 .666 .724 .774 .815 .850 .983 .993 .993 .998 1.000

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TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(11) $T_{\infty}/T_{W} = 4$; Eu = -0.1351; $f_{W} = 0$

 $\frac{\delta^* \sqrt{Re}}{x} = 6.950; \frac{\delta_1 \sqrt{Re}}{x} = 3.109; \frac{\delta_c \sqrt{Re}}{x} = 1.834$

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		•						- NA	
η	f	f'	f"	ſ"	θ	θ'	θ"	u/U∞	ρυ ρ ω Ūω
0 .11.1.1.2.2.2.2.2.3.3.3.4.4.4.5.5.6.6.6.7.7.8.8.8.9.0.8.6.4.2.0.0.0.0.0.0.	0 .000 .000 .000 .001 .002 .004 .006 .009 .017 .022 .034 .042 .050 .070 .092 .119 .260 .304 .351 .402 .515 .640 .707 .850 .707 .850 .1.696 .862 .24.662 .24.662 .2662 .2662 .270 .2662	0 .001 .002 .004 .007 .010 .013 .016 .020 .024 .028 .032 .036 .040 .048 .052 .061 .079 .088 .097 .106 .115 .124 .132 .149 .157 .164 .178 .208 .218 .226 .232 .237 .241 .244 .248 .250 .250 .250 .250	0 .005 .009 .012 .014 .015 .016 .018 .019 .020 .020 .021 .021 .022 .022 .022 .022	0.034 .022 .015 .011 .009 .007 .006 .004 .003 .002 .002 .001 .001 .001 .001 .000 .000	0 .034 .066 .096 .124 .151 .177 .201 .225 .248 .270 .291 .311 .3351 .370 .389 .429 .459 .459 .553 .582 .609 .636 .685 .708 .751 .771 .789 .840 .868 .893 .914 .932 .947 .959 .977 .984 .982 .997 .998 .999 1.000 1.000 1.000 1.000 1.000 1.000	0.179 .165 .154 .145 .137 .131 .125 .120 .116 .112 .108 .105 .102 .099 .096 .094 .092 .087 .073 .070 .065 .053 .051 .048 .046 .043 .038 .038 .029 .024 .011 .009 .007 .005 .001 .001 .000 .000 .000	-0.082063050041035030026023019017016014013012011010009008008007006	0 .001 .002 .005 .009 .014 .020 .026 .033 .041 .050 .069 .079 .102 .114 .139 .167 .196 .228 .291 .325 .325 .325 .325 .325 .325 .325 .325	0 .002 .008 .017 .027 .039 .052 .065 .080 .094 .110 .126 .142 .159 .176 .193 .210 .245 .281 .316 .352 .388 .424 .459 .494 .528 .562 .595 .626 .657 .686 .714 .790 .834 .871 .903 .929 .965 .977 .986 .992 .965 .977 .986 .992 .965 .977 .986 .992 .965 .977 .986 .992 .965 .977 .986 .992 .965 .977 .986 .992 .965 .977 .986 .992 .965 .977 .986 .992 .965 .977 .986 .992 .965 .977 .986 .992 .996 .900 1.000 1.000 1.000 1.000 1.000 1.000 1.000 1.000

(12)
$$T_{\infty}/T_{W} = 4$$
; Eu = -0.09; $f_{W} = 0$

$$\frac{\delta^* \sqrt{Re}}{x}$$
 = 2.297; $\frac{\delta_1 \sqrt{Re}}{x}$ = 2.719; $\frac{\delta_0 \sqrt{Re}}{x}$ = 2.595

		х		x		, <u>x</u>	- 2.000	NA.	CA
η	f	f¹	f"	ſ f"I	θ	θ'	θ"	u/U∞	ρω <mark>U ∞</mark>
0 1234-56-7 111-4-68-0-24-68-2-60-48-2-60-86-4-2-0-8-6-8	.003	0 .018 .032 .045 .056 .066 .075 .090 .097 .104 .109 .125 .134 .149 .156 .168 .174 .183 .192 .206 .233 .236 .239 .244 .248 .249 .250 .250	0.193 .160 .137 .118 .093 .084 .077 .070 .065 .060 .056 .053 .042 .038 .020 .018 .016 .014 .011 .010 .009 .008 .006 .005 .000 .000 .000 .000 .000	-0.391277206158124100082058050043037033021018015011010009007006005005004003003003003003003000000000000000000000000	0 .026 .050 .073 .094 .1156 .136 .155 .174 .227 .243 .275 .336 .390 .416 .510 .552 .628 .6662 .750 .776 .799 .820 .840 .858 .916 .926 .936 .936 .936 .936 .936 .936 .936 .93	0.264 .248 .235 .223 .213 .205 .197 .199 .164 .156 .143 .137 .131 .126 .143 .137 .109 .102 .095 .088 .082 .076 .056 .051 .047 .023 .023 .018 .010 .000 .000 .000 .000	-0.179146123105091081072059059054050046043038034030028028029019018017016015014013012011010010009008007006005001000	0 .019 .0375 .0755 .0896 .1288 .1849	0 .070 .130 .254 .300 .3369 .414 .459 .5567 .5974 .649 .535 .768 .849 .9214 .535 .768 .8798 .8798 .9957 .880 .9957 .998 .998 .9998 .

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TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(13) $T_{\infty}/T_{W} = 4$; Eu = -0.05; $f_{W} = 0$ $\frac{\delta^{*}\sqrt{Re}}{x} = 1.810; \frac{\delta_{1}\sqrt{Re}}{x} = 2.582; \frac{\delta_{C}/Re}{x} = 2.651$

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η	f	f¹	f"	f"I	θ	θ,	θ"	u∕U∞	ρυ ρωθω
0 123468024682604826086420866642086666666666	0 .001 .004 .009 .015 .022 .031 .040 .050 .062 .086 .099 .127 .188 .2216 .292 .367 .446 .528 .613 .700 .881 .163 .1.256 .1.454 .1.454 .1.452 .1.948 .1.452 .1.948 .1.452 .1.948 .1.452 .1.948 .1.452 .1.948 .1.452 .1.948 .1.452 .1.948	0 .022 .040 .055 .068 .079 .090 .107 .114 .121 .128 .134 .144 .154 .162 .169 .176 .188 .193 .202 .209 .216 .221 .226 .230 .234 .237 .239 .242 .245 .245 .245 .245 .245 .249 .250	0.240 .196 .164 .140 .122 .108 .096 .087 .079 .035 .032 .029 .026 .024 .021 .018 .015 .013 .013 .010 .008 .007 .006 .005 .001 .000 .000	-0.529365271205161129054046041032026015013012010008007006005005004003003003002002002002002002002000000	0 027 .052 .076 .099 .121 .162 .182 .219 .237 .254 .287 .219 .237 .254 .287 .219 .349 .349 .349 .349 .349 .349 .349 .34	0.279 .261 .247 .234 .214 .206 .199 .186 .180 .175 .170 .162 .154 .141 .135 .130 .125 .120 .111 .103 .096 .089 .076 .070 .064 .059 .054 .049 .045 .041 .033 .026 .021 .016 .012 .009 .006 .005 .003 .001 .001 .000 .000	-0.199160134114099087070063058053049046041033026024023021019016015014015014015011010009008006005001001001001001001000000000000	0 .024 .046 .067 .088 .108 .128 .129 .236 .269 .233 .361 .339 .418 .445 .471 .561 .561 .561 .555 .783 .810 .855 .855 .855 .855 .855 .855 .855 .85	0 .087 .158 .219 .272 .318 .358 .428 .458 .458 .511 .5357 .614 .648 .750 .771 .806 .837 .925 .947 .958 .964 .984 .996 .998 .998 .998 .998 .998

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TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(14)
$$T_{\infty}/T_{W} = 4$$
; Eu = 0; $f_{W} = 0$

$$\frac{\delta^{*}\sqrt{Re}}{x} = 1.428; \frac{\delta_{1}\sqrt{Re}}{x} = 2.457; \frac{\delta_{C}\sqrt{Re}}{x} = 2.663$$

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			61	f"	f'''	θ	θ'	θ"	u/U _∞	ρu
	η	f	f'	I	1	8		0	u/U _∞	$\rho_{\infty}U_{\infty}$
	0	0	0	0.287	-0.687	0	0.295	-0.222	0	0
١	.1	.001	.026	.230	471	.028	.275	179	.028	.103
1	.2	.005	.047	.191	340	.055	.259	147	.054	.186
1	.3	.011	.064	.161	256	.080	.245	125	.080	.256
-	•4	.018	.079	.138	198	.104	.234	108 094	.104	.316
-	.5	.026	.092	.120	156 127	.127	.215	084	.149	.413
	•6 •7	.036	.113	.095	104	.170	.207	075	.171	.453
	.8	.059	.122	.085	087	.191	.200	068	.192	.489
-	.9	.071	.130	.077	074	.210	.193	062	.213	.522
	1.0	.085	.138	.070	063	.229	.187	057	.233	.551
-	1.1	.099	.144	.064	055	.248	.182	052	.252	.578
	1.2	.114	.151	.059	048	.266	.177	049	.271	.603
	1.3	.129	.156	.055	042	.283	.172	046	.289	.626
	1.4	.145	.162	.051	037	.300	.167	043	.307	.647
Ì	1.6	.178	.171	.044	030 025	.333	.152	035	.375	.718
	2.0	.250	.187	.034	021	.393	.145	032	.407	.747
١	2.2	.288	.193	.030	018	.422	.139	030	.438	.773
	2.4	.327	.199	.027	015	.449	.133	028	.467	.796
	2.6	.368	.204	.024	013	.475	.128	026	.495	.817
	2.8	.409	.209	.022	011	.500	.122	025	.522	.835
	3.2	.494	.217	.018	009	.547	.113	023	.573	.867
	3.6	.582	.223	.015	007	.591	.104	021 019	.619	.889
	$\frac{4.0}{4.4}$.765	.229	.012	006	.631	.089	018	.700	.932
	4.8	.859	.237	.008	004	.702	.082	017	.735	.947
	5.2	.954	.240	.007	004	.733	.075	016	.767	.959
	5.6	1.050	.242	.005	003	.762	.069	015	.796	.968
	6.0	1.148	.244	.004	002	.788	.063	014	.821	.976
	6.4	1.246	.246	.003	002	.812	.057	013	.844	.983
	6.8	1.344	.247	•003	002	.834	.052	013	.865 .883	.988
	7.2 8.0	1.443	.248	.002	002 001	.854 .888	.047	012 010	.912	.996
	8.8	1.642	.249	.000	001	.915	.030	009	.934	.998
	9.6	2.040	.249	.000	.000	.937	.024	007	.950	.998
	10.4	2.240	1			.954	.019	006	.963	1
	11.2	2.440				.967	.014	005	.973	
	12.0	2.640				.977	.011	004	.980	
	12.8	2.840				.985	.008	003	.986	1.
	13.6	3.040				.990	.006	002	.990	1
	14.4	3.240				.994	.003	001	.995	
	16.0	3.640				.998	.002	001	.996	
	16.8	3.840				.999	.001	001	.997	
	17.6	4.040				1.000	.001	.000	.998	
	18.4	4.240		'		1.000	.000	· .000	.998	
	19.2	4.440	1 .	1		1.000	.000	.000	.998	
	20.0	4.640				1.000	.000	.000	.998	
	20.8	4.840	<u> </u>	1		1.000	.000	.000	.998	

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH PORCUS WALL - Continued

(15)
$$T_{\infty}/T_{W} = 4$$
; Eu = 0.5; $f_{W} = 0$

$$\frac{\delta^* \sqrt{Re}}{x} = 0.588; \frac{\delta_1 \sqrt{Re}}{x} = 1.887; \frac{\delta_C \sqrt{Re}}{x} = 2.344$$

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				,				NA NA	مرسه کل
η	f	ſ¹.	f"	f"	θ	θ!	θ"	u/U _∞	ρυ ρωŪω
0 123456789012346802468024660482604826112222233333444555666677888890112234420866642086664208666420866642086664208666420866642086664208666420866642086664208666420866642086666666666	0 .002 .009 .018 .030 .044 .059 .076 .094 .113 .153 .174 .196 .218 .263 .310 .357 .406 .454 .553 .653 .753 .1.553 .1.553 .1.553 .1.553 .1.553 .1.553 .1.553 .1.553 .2.5553 .2.5553 .2.5553 .2.5553 .2.5553 .2.5553 .2.5553 .2.5553 .2.5555 .2.5555 .2.5555 .2.5555 .2.5555 .2.5555 .2.5555 .2.5555 .2.5555 .2.5555 .2.5555 .2.5555 .2.5555 .2.5555 .2.5555 .2.5555 .2.5	0 .046 .080 .108 .129 .147 .161 .174 .184 .193 .213 .236 .243 .246 .249 .250 .250 .250	0.537 .394 .302 .239 .194 .160 .134 .113 .097 .083 .072 .062 .054 .047 .019 .014 .010 .007 .005 .003 .001	-1.810 -1.117748529387295230184150123087074064055042032026016014011009006	0 .037 .071 .103 .133 .161 .188 .214 .239 .263 .286 .308 .350 .370 .408 .448 .510 .541 .570 .597 .623 .648 .671 .693 .734 .770 .803 .734 .770 .803 .856 .879 .999 .999 .999 .999 .999 .999 .999	0.388 .354 .329 .308 .291 .276 .263 .252 .242 .233 .225 .217 .210 .203 .197 .185 .175 .166 .157 .149 .141 .133 .126 .120 .086 .067 .059 .052 .045 .039 .024 .021 .003 .003 .001 .000 .000 .000	-0.384 -0.289 -0.289 -0.187 -0.157 -0.136 -0.119 -0.064 -0.064 -0.064 -0.064 -0.040 -0.038 -0.036 -0.036 -0.031 -0.030 -0.027 -0.025 -0.025 -0.021 -0.016 -0.014 -0.013 -0.016 -0.006 -0.006 -0.006 -0.006 -0.000 -0.000 -0.000 -0.000	0 .051 .098 .141 .180 .218 .252 .285 .316 .345 .373 .424 .448 .470 .5584 .616 .644 .670 .694 .716 .735 .770 .828 .873 .999 .924 .936 .947 .999 .999 .999 .999 .999 .999 .999	0 .184 .323 .432 .516 .586 .645 .694 .736 .772 .830 .853 .874 .891 .920 .973 .983 .995 .998 1.000

224C

13.60 3.293 14.40 3.493

15.20 3.693

(16)
$$T_{\infty}/T_{W} = 4$$
; Eu = 1.0; $f_{W} = 0$

$$\frac{6^* \sqrt{Re}}{x} = 0.427; \frac{6_1 \sqrt{Re}}{x} = 1.615; \frac{6_c \sqrt{Re}}{x} = 2.075$$

ρu f" ſ ſ١ θ" η ſ θ 01 u/U_∞ PaUa 0.685 0 -2.764 0.453 -0.523 .05 .001 .031 -2.062 .566 .022 .429 -.440 .033 .124 .10 .003 .057 .476 -1.586 .043 -:378 .228 .064 .15 .20 .006 .079 .405 -1.251 .063 -.329 .391 .094 .316 .011 -.291 .098 .349 -1.007 .082 .376 .122 .391 .25 .016 .114 .303 -.824 .101 .362 -.260 .148 .456 .30 .022 .128 .266 -.684 .118 .349 -.234 .174 .514 .141 .152 .35 .029 .234 -.575 .338 -.213 .207 .564 .40 .036 .208 -.490 .152 .328 -.195 .221 .608 .45 .50 .55 .044 .162 .185 -.420 .168 .319 -.179 .243 .647 .170 .166 -.363 .184 .310 -.166 .265 .682 .061 -.155 .178 .149 -.316 .200 .302 .285 .713 -.277 -.245 .60 .070 .185 .134 .215 .305 .295 -.145 .742 .65 .080 .192 .121 .288 .229 -.137 .324 .767 .70 .089 .198 -.217 .243 .109 .281 -.128 .342 .790 .75 .099 .203 .099 -.193 .257 .275 -.122 .359 .811 .110 .120 .131 .208 .212 .216 .090 .80 -.173 .271 .269 -.116 .376 .830 .85 .082 -.155 .284 .263 -.110 .392 .847 .90 .074 -.140 .297 .258 -.105 .408 .863 .95 .142 .219 .068 -.126 .310 .253 -.101 .423 .877 .153 .222 .062 -.115 .323 -.096 .248 .438 .890 .347 .371 .393 1.10 .175 .228 .051 -.095 .238 -.089 .466 .912 .230 1.20 .198 .233 .043 -.079 -.083 .492 .931 .237 1.30 .222 .035 -.067 -.078 .516 .947 1.40 .246 .240 .029 -.057 .415 .214 -.074 .539 .960 1.50 .270 .243 .024 -.049 .436 .207 -.070 .560 .970 .294 1.60 .245 .019 -.042 .456 .200 -.066 .580 .979 1.70 .246 .016 -.036 -.063 .476 .194 .599 .986 .343 .248 1.80 .012 -.031 .495 .187 -.061 .616 .992 2.00 .007 -.024 .531 .176 -.056 648 .999 2.20 .443 .250 .676 .002 -.018 .565 .165 -.052 1.000 2.40 .493 .250 -.001 -.014 .597 .155 -.049 .700 1.000 2.60 .543 .627 .14.5 -.046 2.80 .593 -.044 -.041 .656 .136 .744 3.00 .643 .682 .128 .764 .707 3.20 .693 .120 -.039 .782 3.40 .743 .730 .752 .112 -.037 .800 .793 3.60 -.035 .105 .816 3.80 .843 .772 .098 -.034 .832 .893 4.00 .791 .092 -.032 .846 .993 4.40 .825 .079 -.029 .872 4.80 1.093 .855 .068 -.026 .894 5.20 1.193 .880 .059 -.023 -.021 .913 .050 5.60 1.293 .902 .930 6.00 1.393 .920 -.018 .042 .943 6.40 1.493 .936 .035 -.016 .955 6.80 1.593 .949 .029 -.014 .965 7.20 1.693 .959 .024 -.012 .973 7.60 1.793 .968 .020 -.010 .979 8.00 1.893 8.40 1.993 .975 .016 -.009 .985 .981 .013 -.007 .989 8.80 2.093 -.006 .985 .010 .992 9.60 2.293 .992 .006 -.004 .997 10.40 2.493 .996 .004 -.002 .999 11.20 2.693 .998 .002 -.002 1.000 12.00 2.893 .999 -.001 .001 1.000 12.80 3.093 1.000 .001 .000 1.000

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TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(17) $T_{\infty}/T_{W} = 1$; Eu = -0.0418; $f_{W} = -0.5$ $\frac{\delta^{*}\sqrt{Re}}{x} = 4.272$; $\frac{\delta_{i}\sqrt{Re}}{x} = 0.954$; $\frac{\delta_{c}\sqrt{Re}}{x} = 0.807$

η	f	f'	f"	f""	θ	θ'	θ"
0	-0.500 500 496 487 469 437 389 319 225 101 .055 .246 .474 .736 1.032 1.356 1.704 2.070 2.450 2.838 3.232 3.630 4.428 4.828 5.628 4.428 4.828 7.628 8.028		0 .018 .037 .058 .081 .106 .132 .158 .182 .204 .219 .225 .221 .206 .181 .149 .115 .082 .055 .034 .019 .010 .005 .002 .001	0.042 .046 .051 .055 .060 .064 .058 .047 .028 .004 051 073 085 076 061 044 029 017 010 005 002	.994 .997 .998 .999		0.017 .018 .020 .020 .021 .021 .020 .017 .012 .006 003 014 025 036 045 050 051 048 042 034 026 018 004 002 001 001 001 000 000 000

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(18)
$$T_{\infty}/T_{W} = 1$$
; Eu = 0; $f_{W} = -0.5$ NACA
$$\frac{\delta^{*}\sqrt{Re}}{x} = 2.459; \frac{\delta_{1}\sqrt{Re}}{x} = 0.827; \frac{\delta_{C}\sqrt{Re}}{x} = 0.973$$

η	f	f†	f"	f"	θ	θ 1	θ"
0 .4 8 2 1.6 0 2.4 8 2 3.6 0 4 4 8 2 5 6 0 6 .4 8 7 7 6 0 8 8 8 9 .6 10 .4 10 .8	-0.500 486 444 369 260 114 .070 .293 .553 .170 1.518 1.884 2.264 2.652 3.443 3.842 4.641 5.441 5.441 5.441 7.441 7.441 7.841	0 .070 .146 .228 .318 .412 .509 .604 .774 .841 .895 .933 .962 .980 .990 .995 .998 1.000 1.000	0.165 .182 .200 .217 .231 .240 .241 .232 .214 .186 .152 .116 .083 .055 .034 .019 .010 .005 .002 .001	0.041 .044 .040 .030 .014 008 059 079 089 088 078 062 044 029 017 009 001	0 .069 .142 .221 .303 .389 .476 .562 .644 .720 .788 .846 .892 .928 .972 .984 .991 .995 .998 .999 1.000 1.000 1.000 1.000 1.000	0.166 .178 .190 .201 .210 .216 .217 .212 .200 .181 .157 .130 .103 .077 .054 .036 .023 .014 .008 .004 .002 .001 .000 .000 .000	0.029 .030 .030 .026 .019 .009 005 022 039 064 069 068 061 050 068 061 050 039 028 019 002 007 004 002 000 000

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(19)
$$T_{\infty}/T_{W} = 1$$
; Eu = 0.5; $f_{W} = -0.5$

$$\frac{\delta^{*}\sqrt{Re}}{x} = 1.033$$
; $\frac{\delta_{1}\sqrt{Re}}{x} = 0.444$; $\frac{\delta_{C}\sqrt{Re}}{x} = 0.994$

π f f f f f f g g 0 -0.500 0 0.697 -0.238 0 0.259 0.068 .4 447 .259 .596 267 .109 .287 .067 .6 384 .373 .542 275 .168 .300 .060 .8 298 .476 .486 278 .229 .311 .049 1.0 194 .567 .431 276 .292 .311 .049 1.2 072 .648 .376 270 .356 .324 .011 1.6 .215 .778 .273 242 .486 .319 036 1.8 .375 .827 .227 222 .548 .309 061 2.0 .545 .868 .185 198 .609 .295 084 2.2 .722 .902 .148
.2 486 .135 .648 254 .053 .273 .070 .4 447 .259 .596 267 .109 .287 .067 .6 384 .373 .542 275 .168 .300 .060 .8 298 .476 .486 278 .229 .311 .049 1.0 194 .567 .431 276 .292 .319 .032 1.2 072 .648 .376 270 .356 .324 .012 1.4 .065 .718 .323 258 .421 .324 011 1.6 .215 .778 .273 242 .486 .319 036 1.8 .3375 .827 .227 .222 .548 .309 .061 2.0 .545 .868 .185 .198 .609 .295 .084 2.2 .722 .902 .148<
7.0 5.467 1.000 .0

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(20)
$$T_{\infty}/T_{W} = 1$$
; Eu = 1; $f_{W} = -0.5$ NACA
$$\frac{\delta^{*}\sqrt{Re}}{x} = 0.783; \frac{\delta_{1}\sqrt{Re}}{x} = 0.345; \frac{\delta_{C}\sqrt{Re}}{x} = 0.918$$

η	f	f'	f"	f"	θ	91	θ"
0 2468024680246802468024	-0.500 481 428 345 236 106 .042 .203 .375 .556 .742 .933 1.128 1.521 1.720 1.919 2.318 2.517 2.917 3.317 3.517 3.517 3.517 4.517 4.517 4.517 4.517 4.517 4.517 5.117	0 .183 .344 .483 .601 .697 .775 .884 .920 .946 .977 .985 .991 .998 .998 .998 .998	0.969 .862 .750 .640 .534 .435 .347 .270 .205 .152 .076 .052 .013 .008 .004 .000 .000	-0.516 551 560 545 513 468 413 295 239 142 105 075 052 034 009 003	0 .061 .126 .194 .267 .341 .418 .493 .566 .636 .700 .758 .858 .918 .951 .987 .995 .995 .995 .999 .999 .999 .999 .99	0.293 .314 .335 .354 .368 .377 .383 .373 .358 .335 .272 .236 .199 .163 .130 .101 .076 .056 .040 .028 .012 .008 .005 .000 .000 .000	0.103 .106 .100 .085 .061 .028 011 053 094 130 159 178 186 135 12 090 070 052 038 026 018 012 008 012 008 005 005 000 .000

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TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(21) $T_{\infty}/T_{W} = 2$; Eu = 0; $f_{W} = -0.5$

 $\frac{\delta^* \sqrt{Re}}{x} = 2.381; \frac{\delta_1 \sqrt{Re}}{x} = 1.605; \frac{\delta_c \sqrt{Re}}{x} = 1.778$

NACA

		X				, .			
								4 8 1 1	ρu
η	· f	f'	f"	f"1	θ	θ1.	θ"	u/U∞	$\rho_{\infty}U_{\infty}$
0	-0.500	0	0.148	-0.027	0	0.160	0.006	0	0 050
.2	497	.029	.142	025	.032	.161	.006	.030	.058
.4	488	.057	.138	024	.065	.163	.006	.061	.114
.6	474	.084	.133	023	.097	164	.005	.092 .124	.220
.8	455	.110	.128	022	.130	.165	.004	.157	.271
1.0	430	.135	.124	022	.163	.165	.001	191	.319
1.2	401	.160	.119	022 022	.196 .229	.166	.000	.225	.366
1.4	367	.183	.115	022	.262	.165	002	.260	.411
1.6	328	.206	.106	022	.295	.165	004	.294	.455
1.8	284	.227	.102	022	.328	.164	006	.330	.496
2.0	237 130	.287	.093	023	.393	.161	010	.400	.574
2.8	008	.322	.083	023	.457	.156	014	.469	.644
3.2	.128	.354	.074	023	.518	.150	018	. 537	.707
3.6	.275	.382	.065	023	.576	.142	-:021	.601	.763
4.0	.432	.406	.056	022	.631	.133	024	.662	.811
4.4	.599	.426	.047	021	.682	.123	026	.717	.853
4.8	.773	.444	.039	017	.729	.112	027	.767	.887
5.2	.953	.458	.032	017	.772	.101	028	.811 .850	.916 .939
5.6	1.139	.469	.026	015	.810	.090	028 027	.882	.957
6.0	1.329	.478	.020	013	.844	.079	026	909	.970
6.4	1.521	.485	.015	011	.873 .899	.058	024	.931	.981
6.8	1.717	.490	.011	007	.920	.049	022	.948	.988
7.2	1.914	.494	.003	004	.952	.033	018	.972	.996
8.0	2.310	.500	.001	002	.974	.021	013	.986	.999
9.6	3.109	.500	.000	001	.986	.012	009	.993	1.000
10.4	3.509				.994	.006	005	.996	
11.2	3.909				.997	.003	003	.998	
12.0	4.309			١,	.999	.001	001	.999	'
12.8	4.709		1		1.000	.000	001	.999	·
13.6	5.109	ļ. ,			1.000	.000	.000	1.000	
14.4	5.509		2 1	1	1.000	.000	.000	1.000	

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(22)
$$T_{\infty}/T_{W} = 2$$
; Eu = 0.5; $f_{W} = -0.5$
 $\frac{\delta^{*}\sqrt{Re}}{x} = 0.877$; $\frac{\delta_{1}\sqrt{Re}}{x} = 1.117$; $\frac{\delta_{C}\sqrt{Re}}{x} = 1.760$

η f f¹ f" r" θ θ¹ θ¹ μ√U∞ ρυ 0 -0.500 0 0.473 -0.365 0 0.229 0.016 0 0 .4 466 .162 .344 323 .046 .232 .011 .178 .325 .6 427 .226 .291 249 .140 .236 .007 .257 .452 .8 376 .279 .244 217 .187 .237 .001 .331 .558 1.0 316 .324 .204 188 .234 .237 .004 .400 .648 1.2 248 .361 .169 162 .282 .235 011 .463 .722 1.4 172 .392 .139 138 .328 .233 017 .520 .783 1.6 091 .417 .113 118 .375 .										
.2	η	f	f	f"	Lut	θ	θ1	θ"	u/U∞	
10.8 4.461 1.000 .000 001 1.000	24.68024680246826048260482604 11.22222233333444.556666777888.8990010.4	491466427376248172091006 .270 .366 .464 .562 .761 .961 1.461 1.661 1.461 1.661 1.261 1.661 1.261 1.661 1.261 1.661 1.261 1.261 1.261 1.261 1.261	.088 .162 .226 .279 .324 .361 .392 .417 .437 .454 .467 .477 .485 .491 .495 .498 .500	.404 .344 .291 .244 .204 .169 .139 .113 .092 .073 .058 .045 .034 .025 .018 .012	323284249217188162138118100084071059049041033027022018	.046 .093 .140 .187 .234 .282 .328 .375 .420 .464 .506 .547 .587 .626 .693 .7751 .800 .885 .915 .936 .915 .936 .997 .999 .999 .999 .999 .999 .999 .99	.232 .235 .236 .237 .235 .233 .229 .223 .217 .210 .201 .192 .183 .172 .162 .151 .141 .130 .120 .081 .065 .050 .038 .028 .021 .015 .010 .007 .004 .003 .002 .001 .000	.014 .011 .007 .001 004 011 017 024 029 035 040 044 047 050 053 05	0 .092 .178 .257 .331 .400 .463 .520 .573 .621 .664 .703 .738 .769 .797 .821 .843 .861 .872 .904 .926 .944 .959 .970 .979 .979 .996 .998 .999 .998 .999 .999 .999 .99	0 .176 .325 .452 .558 .648 .722 .783 .834 .907 .933 .954 .969 .995 .999 .995

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(23)
$$T_{\infty}/T_{W} = 2$$
; Eu = 1.0; $f_{W} = -0.5$
 $\frac{\delta^{*}\sqrt{Re}}{x} = 0.637$; $\frac{\delta_{1}\sqrt{Re}}{x} = 0.968$; $\frac{\delta_{c}\sqrt{Re}}{x} = 1.613$

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								-w/	<i></i>
η	f	f¹	f"	f"	θ	θι	θ ή	u/U _∞	ρυ ρω0 ω
0	-0.500 488	0	0.634 .519	-0.615 542	0	0.253	0.034	0.121	0
.2	456	.208	.418	467	.104	265	.025	.230	.417
.6	406	.283	.332	395	.157	.269	.016	.328	.567
.8	344	.342	.259	328	.211	.271	.006	.414	.684
1.0	270	.388	.200	269	.265	.271	005	.491	.776
1.2	189	.423	.151	218	.319	.269 .265	017 028	.558	.845 .898
1.4	102	.449 .468	.112	175 139	.373 .425	.258	038	.667	.936
1.6	010 .085	.482	.056	110	.476	.249	047	711	.963
2.0	.182	.491	.036	086	.525	239	055	.748	.982
2.2	.281	.497	.021	068	.572	.228	061	780	.993
2.4	.381	.500	.009	053	.616	.215	066	.807	.999
2.6	.481	.500	•000	041	.657	.201	069	.830	1.000
2.8	.581				.696	.187	071	.849	
3.0	.681			1	.732	.173	072	.867	
3.2	.781			100	.765	.158	071	.884	
3.6	.981				.823	.130	068 062	.912	
4.0	1.181		,		.870 .907	.081	054	.954	
4.4	1.381				.935	.061	045	968	
5.2	1.781				.956	.045	037	.979	
5.6	1.981				.971	.032	029	.987	
6.0	2.181				.982	.022	021	.992	
6.4	2.381	ĺ			.989	.015	015	.996	
6.8	2.581				.994	.009	011	.998	
7.2	2.781				.997	.006	007	1.000	
7.6	2.981				.999	.003	004	1.000	
8.0	3.181				1.000	.002	003 001	1.000	
8.4	3.381 3.581		2	: 1	1.000	.001	001	1.000	
9.2	3.781			٠. ا	1.000	.000	001	1.000	
9.6	3.981				1.000	.000	.000	1.000	
10.0	4.181				1.000	.000	•000	1.000	

(24) $T_{\infty}/T_{W} = 4$; Eu = -0.0644; $T_{W} = -0.5$

 $\frac{\delta^* \sqrt{Re}}{x} = 7.219; \frac{\delta_1 \sqrt{Re}}{x} = 3.484; \frac{\delta_c \sqrt{Re}}{x} = 2.620$ NACA

	T	· x	- = /.21	y X	= 3.48	4; X	- = 2.6	20	NACA
, n	f	f'	f"	f"	θ	θ'	θ"	u/U _∞	<u>ρυ</u> ρ _ω υ _ω
0 .46.04.82.604.82.08.64.208.208.64.208.64.208.64.208.64.208.64.208.64.208.64.208.64.208.64.208.64.208.64.208.64.208.64.2	500 500 499 498 494 492 485 475 462 446	0 .000 .001 .002 .004 .006 .009 .011 .014 .021 .028 .036 .044 .053 .062 .071 .080 .089 .098 .117 .126 .135 .152 .168 .182 .195 .206 .216 .224 .231 .236 .241 .244 .246 .248 .249 .250 .250 .250 .250 .250 .250 .250 .250	0 .003 .006 .008 .010 .011 .013 .014 .015 .017 .019 .020 .021 .022 .023 .023 .023 .023 .023 .023 .023	0.016 .014 .012 .010 .009 .008 .007 .006 .005 .004 .003 .002 .001 .001 .000 .000 .000 .000 .001 .001 .001 .002 .002	0 .016 .032 .048 .063 .079 .094 .110 .125 .187 .218 .278 .308 .338 .368 .397 .426 .454 .512 .540 .593 .644 .692 .736 .777 .814 .847 .847 .900 .921 .939 .955 .995 .995 .995 .995 .995 .995	0.080 .079 .079 .078 .078 .078 .077 .077 .077 .076 .076 .076 .076 .075 .075 .075 .074 .073 .072 .071 .070 .068 .065 .062 .058 .053 .048 .033 .033 .029 .024 .020 .016 .013 .010 .000 .000 .000 .000	-0.003003002001001001001001001001001001001002002002003004006	0 .000 .001 .003 .005 .008 .011 .015 .020 .031 .044 .060 .077 .119 .143 .168 .196 .225 .255 .287 .319 .353 .422 .491 .560 .626 .687 .744 .794 .838 .875 .906 .931 .950 .965 .976 .994 .999 .998 .999 .998 .999 .998 .999 .998 .999 .998 .999 .998 .999 .990 .900 .90	0 .001 .005 .010 .017 .025 .035 .046 .058 .084 .113 .144 .177 .211 .247 .283 .320 .357 .394 .431 .468 .539 .607 .670 .728 .780 .825 .864 .897 .924 .946 .995 .995 .995 .998 .995 .998 .995 .998 .995 .998 .995 .998 .990 .995 .995

(25)
$$T_{\infty}/T_{W} = 4$$
; Eu = 0; $f_{W} = -0.5$
 $\frac{\delta^{*}\sqrt{Re}}{x} = 2.460$; $\frac{\delta_{1}\sqrt{Re}}{x} = 3.100$; $\frac{\delta_{C}\sqrt{Re}}{x} = 3.236$

(26) $T_{\infty}/T_{W} = 4$; Eu = 0.5; $f_{W} = -0.5$

 $\frac{\delta^* \sqrt{Re}}{x} = 0.773; \frac{\delta_1 \sqrt{Re}}{x} = 2.486; \frac{\delta_c \sqrt{Re}}{x} = 3.123$

NACA

		х		, x		, х		MA	CA
η	f	f	f"	ſ"¹	θ	81	θ"	ս/Մ∞	ρυ ΡωŪ∞
0 24.68 1.24.68 1.222.48 2.60 4.82.60 4.82.60 4.82.60 8.64.20 8.64.20 8.64.20 12.86.42 13.46.82 14.46.83 14.46.	-0.500494478475426392355229184137089 .107 .207 .307 .407 .507 .607 .707 .807 .107 1.207 1.107 1.507 1.707 1.907 2.107 2.307 2.507 2.707 2.507 2.707 3.507 3.507 3.507 3.507 3.507 3.707 3.907 4.107	0 .057 .099 .132 .157 .177 .193 .206 .216 .224 .231 .236 .249 .250 .250	0.331 .243 .185 .143 .112 .089 .071 .057 .046 .036 .029 .023 .018 .010 .005 .001 002	-0.539352245177132101079063050041033028023016012008006	0 .039 .077 .113 .217 .250 .282 .313 .373 .402 .457 .510 .559 .604 .723 .756 .787 .814 .839 .861 .881 .927 .949 .949 .996 .997 .998 .999 .999 .999 .999 .999 .999	0.201 .192 .185 .180 .175 .170 .166 .162 .158 .154 .150 .146 .143 .135 .127 .119 .111 .103 .095 .087 .079 .072 .065 .059 .052 .046 .041 .023 .017 .012 .008 .005 .003 .000 .000 .000	-0.050037031026023020	0 .064 .122 .177 .228 .275 .319 .360 .399 .435 .469 .500 .530 .583 .629 .669 .703 .735 .765 .792 .817 .840 .861 .879 .896 .911 .924 .946 .946 .946 .974 .989 .999 .999 .999 .999 .999 .999 .99	0 .228 .398 .528 .629 .709 .772 .824 .897 .923 .944 .960 .983 .995 1.000

(27)
$$T_{\infty}/T_{W} = 4$$
; Eu = 1.0; $f_{W} = -0.5$
 $\frac{\delta^{*}\sqrt{Re}}{x} = 0.553$; $\frac{\delta_{1}\sqrt{Re}}{x} = 2.235$; $\frac{\delta_{C}\sqrt{Re}}{x} = 2.861$

NACA

									-
η	f	f'	f"	f"	θ.	θ,	θ"	ս/Մ∞	pu pællæ
0 .46802482604826048642086420864455566667788889042122334451668	-0.500492473445410371328235138088038062162362462662762662762162762162762162762162762	0 .071 .122 .159 .186 .206 .221 .239 .244 .247 .249 .250	0.422 .298 .216 .156 .085 .062 .031 .021 .006 .001	-0.768499346242176131099059046036029023	0 .042 .083 .123 .161 .199 .236 .272 .308 .342 .376 .409 .440 .501 .557 .610 .658 .702 .742 .778 .811 .839 .865 .923 .937 .999 .974 .991 .995 .997 .999 1.000 1.000 1.000 1.000 1.000 1.000	0.213 .206 .201 .196 .192 .188 .183 .179 .175 .170 .166 .161 .156 .146 .126 .115 .095 .086 .076 .068 .059 .052 .045 .033 .027 .023 .015 .006 .004 .002 .001 .000 .000	-0.041031025021021021023023024025026026026026025024023021020018017015014012011008005004002001000000000000	0 .080 .152 .218 .276 .329 .377 .420 .460 .495 .555 .580 .626 .668 .707 .743 .776 .804 .838 .8915 .932 .942 .952 .970 .988 .993 .998 .998 .999 .999 .999 .999	0 .285 .488 .637 .745 .825 .883 .926 .976 .990 .997 1.000

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS
IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES,
PRESSURE GRADIENT IN MAIN STREAM, AND FLOW
THROUGH POROUS WALL - Continued

(28) $T_{\infty}/T_{W} = 1$; Eu = -0.0072; $f_{W} = -1.0$

 $\frac{\delta^* \sqrt{Re}}{x} = 6.398; \frac{\delta_i \sqrt{Re}}{x} = 1.116; \frac{\delta_c \sqrt{Re}}{x} = 1.072$

η	f	f'	f"	f"	θ	θ1	θ"
0 .50 1.00 1.50 2.50 3.50 3.50 4.50 5.50 5.50 5.50 7.25 7.50 5.50 7.25 9.00 5.50 9.25 9.00 10.25 10.25 11.50	-1.000 -1.000999995987951916864791556381160030 .112 .267 .434 .806 1.007 1.218 1.438 1.664 1.896 2.133 2.374 2.617 2.863 3.110 3.357 3.606 3.855 4.104 4.354 4.603 4.853 5.102 6.602 6.852 7.102	0 .001 .004 .011 .021 .035 .056 .085 .172 .233 .308 .395 .492 .543 .646 .695 .742 .786 .862 .8918 .918 .918 .918 .918 .918 .918 .91	0 .004 .009 .016 .024 .035 .049 .066 .110 .136 .162 .186 .202 .194 .183 .168 .151 .132 .112 .093 .075 .059 .044 .033 .023 .016 .011 .007 .004 .003 .002 .001 .000 .000 .000	0.007 .009 .012 .015 .019 .024 .030 .037 .044 .050 .053 .051 .022 .008 073 078 078 079 076 070 062 070 062 052 042 033 013 013 008 001 001 000	0 .014 .030 .049 .072 .100 .132 .170 .215 .267 .327 .394 .468 .547 .587 .628 .668 .707 .745 .781 .845 .873 .918 .936 .951 .963 .972 .980 .999 .999 .999 .999 .999 .999 .999	0.025 .030 .036 .042 .050 .060 .070 .083 .097 .1127 .142 .154 .161 .162 .159 .155 .148 .139 .128 .116 .091 .078 .065 .026 .020 .011 .000 .000 .000 .000 .000 .000	0.009 .010 .012 .015 .017 .020 .023 .026 .029 .031 .030 .027 .000 .002006015023039045049052051048044039044039044039044039044039044039044039044011008006004003001001001000000000000000

(29) $T_{\infty}/T_{W} = 1$; Eu = 0; $f_{W} = -1.0$ $\frac{\delta \sqrt[8]{Re}}{x} = 4.396$; $\frac{\delta_{1}/Re}{x} = 1.073$; $\frac{\delta_{C}/Re}{x} = 1.147$

		- in 1					
η	f	f†	f"	f"1	θ	θ!	θ"
0 .4680246802468024680246802468024680246802	-1.000 9997 9987 9987 9987 9956 9956 9956 9956 9958 9958 7273 4808 76154 4808 76154 4808 76154 1323	0 .016 .025 .046 .025 .046 .025 .046 .025 .046 .026 .035 .046 .026 .035 .046 .026 .035 .046 .026 .036 .036 .036 .036 .036 .036 .036 .03	0.036 .039 .043 .048 .053 .058 .064 .071 .078 .085 .102 .111 .131 .141 .151 .161 .171 .180 .188 .195 .204 .205 .202 .196 .188 .178 .166 .152 .137 .106 .090 .076 .050 .040 .018 .018 .018 .018 .018 .019 .018 .019 .018 .019 .019 .019 .019 .019 .019 .019 .019	0.018 .020 .022 .024 .026 .028 .031 .034 .035 .047 .049 .050 .051 .051 .050 .044 .032 .023 .013 .002 -010 -022 -016 -073 -075 -075 -075 -075 -075 -075 -075 -075	0 .011 .022 .034 .048 .062 .077 .093 .110 .129 .149 .170 .192 .216 .221 .324 .354 .386 .419 .452 .557 .592 .628 .628 .760 .789 .817 .845 .923 .937 .950 .968 .975 .989 .999 .999 .999 .999 .999 .999 .99	0.052 .055 .059 .064 .068 .073 .078 .089 .096 .102 .115 .129 .142 .149 .155 .161 .176 .177 .176 .177 .176 .174 .171 .166 .170 .174 .171 .166 .1099 .088 .099 .096 .000 .000 .000 .000 .000 .000	0.018 0.019 0.021 0.024 0.025 0.027 0.028 0.029 0.031 0.032 0.035 0.034 0.034 0.034 0.034 0.034 0.031 0.006 0.011 0.006 0.057 0.058 0.057

(30)
$$T_w/T_w = 1$$
; Eu = 0.05; $f_w = -1.0$

$$\frac{5 \sqrt[4]{Re}}{x} = 2.796$$
; $\frac{\delta_1/Re}{x} = 0.911$; $\frac{\delta_c/Re}{x} = 1.241$

η	f	f¹	f"	f"	θ	θ1	θ"
0 111122222333333444445555556666667777788888899999999999999999	988 974 953 925 891	1 -	0.141 .146 .152 .158 .164 .171 .178 .184 .191 .198 .204 .219 .214 .210 .202 .193 .182 .168 .154 .138 .122 .106 .090 .076 .062 .050 .040 .031 .023 .017 .013 .009 .001 .000 .000 .000	0.024 .029 .031 .033 .034 .035 .035 .034 .019 .011 .002 008 019 052 070 062 070 064 071 064 071 064 071 064 071 064 071 064 071 064 071 064 071 008 019 008 019 008 019 008 019 008 019 008 019 008 019 008 019 008 019 008 019 008 019 008 019 008 019 008 019 008 019 008 00	018 038 059 082 106 132 160 190 221 254 289 326 364 403 443 443 448 525	0.088 .095 .102 .110 .118 .126 .135 .144 .153 .162 .170 .178 .186 .193 .205 .205 .205 .205 .205 .205 .205 .197 .189 .180 .170 .158 .144 .131 .117 .103 .089 .076 .064 .036 .028 .022 .017 .013 .000 .000 .000 .000 .000 .000 .000	0.032 .037 .039 .041 .043 .044 .045 .044 .045 .017 .009001011021050063067070069066062057070069068014011031014011031041011009066062057070070069066062057070070069066062051045033014011009006005004002001000

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(31)
$$T_{\infty}/T_{W} = 1$$
; Eu = 0.15; $f_{W} = -1.0$

$$\frac{\delta^{*}\sqrt{Re}}{x} = 2.008; \frac{\delta_{1}\sqrt{Re}}{x} = 0.750; \frac{\delta_{c}\sqrt{Re}}{x} = 1.280$$

η	f	f†	f"	f"	θ	θ!	θ"
0 1111122222333333444445555556666667777778888888999990	-1.000 995 978 978 951 864 733 864 7559 3220 3220 323	0 .054 .109 .163 .218 .273 .382 .436 .488 .540 .590 .638 .684 .727 .767 .864 .933 .948 .961 .971 .985 .995 .995 .999 .999 1.000 1.000 1.000 1.000 1.000 1.000 1.000 1.000	0.270 271 272 273 274 274 273 270 267 262 255 246 235 222 208 .192 .174 .156 .138 .119 .085 .070 .056 .044 .026 .019 .010 .007 .005 .003 .002 .001 .000 .000 .000	0.005 .006 .005 .004 .001 003 008 014 022 030 059 069 092 090 086 080 072 086 080 072 030 037 030 013 013 009 001 003 001 000	0 .024 .049 .076 .106 .138 .172 .209 .248 .289 .376 .422 .468 .515 .561 .652 .428 .515 .561 .672 .652 .428 .772 .838 .866 .890 .945 .945 .945 .945 .945 .945 .995 .995	0.113 .122 .132 .143 .154 .166 .177 .189 .299 .218 .225 .230 .233 .234 .231 .226 .218 .208 .195 .148 .195 .148 .195 .045 .045 .045 .028 .028 .028 .028 .028 .028 .028 .029 .028 .029 .028 .029 .028 .029 .028 .029 .029 .029 .029 .029 .029 .029 .029	0.045 .049 .052 .055 .057 .058 .057 .056 .052 .047 .040 .031 .020 .008 019 033 046 058 068 076 082 085 085 085 083 079 073 066 059 073 014 011 023 014 011 004 011 004 004 005 011 006 -

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(32)
$$T_{\infty}/T_{W} = 1$$
; Eu = 0.5; $f_{W} = -1.0$

$$\frac{\delta^{*}\sqrt{Re}}{x} = 1.252$$
; $\frac{\delta_{1}\sqrt{Re}}{x} = 0.524$; $\frac{\delta_{C}\sqrt{Re}}{x} = 1.269$

η	f	f'	f"	f"	θ	. 01	θ"
0 .468024680246802468024680246802468024	-1.000 990 958 958 958 958 959 6533 259 1056 .259 1056 .259 1056 .203 .31.1548 1.3548 1.344 2.344 2.344 2.344 2.344 2.344 2.344 2.344 2.344 2.344 2.344 2.344 3.345 3.345 3.345 3.345 3.345 3.348 4.348 4.348 4.348 4.348 4.348 4.348 5.348 5.348 6.148 5.348 6.148	0 .105 .205 .300 .390 .473 .550 .621 .684 .741 .790 .833 .869 .999 .910 .987 .995 .995 .995 .998 .999 .995 .998 .999 .900 .000 1.000 1.000 1.000	0.534 .513 .489 .462 .433 .402 .369 .334 .299 .264 .230 .197 .166 .137 .111 .088 .068 .052 .039 .028 .020 .014 .010 .007 .005 .002 .002 .002 .001 .001	-0.099114127140152161169174176174170162151137122106089074059046035026018013009006001001000000	0 .029 .062 .098 .138 .181 .228 .331 .386 .443 .500 .557 .613 .666 .716 .763 .804 .841 .901 .924 .942 .957 .969 .942 .957 .969 .999 .999 .999 .999 .999 .999 .99	0.139 .155 .171 .189 .207 .225 .242 .258 .271 .286 .287 .260 .242 .220 .197 .173 .149 .125 .103 .083 .066 .051 .038 .021 .015 .010 .007 .005 .001 .000 .000 .000 .000	0.073 .080 .086 .090 .091 .089 .083 .072 .057 .038 .016 008 058 098 120 123 121 115 105 094 081 068 055 044 068 019 014 010 014 010 007 001 001 001 001 000 .000 .000

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(33)
$$T_{\infty}/T_{W} = 1$$
; Eu = 1.0; $f_{W} = -1.0$

$$\frac{\delta^{*}\sqrt{Re}}{x} = 0.945$$
; $\frac{\delta_{1}\sqrt{Re}}{x} = 0.405$; $\frac{\delta_{C}\sqrt{Re}}{x} = 1.208$

η	f	f'	f"	f"	θ	θ1	θ"
0 11111122222233333334444455555566666677	-1.000 98244200 9878700 9878700 1.0000 1.00000 1.00000 1.00000 1.00000 1.0000 1.0000 1.0000 1.00000 1.00000 1.000	0 .146 .281 .403 .512 .607 .690 .759 .816 .900 .928 .950 .966 .977 .985 .991 .999 .999 .999 .999 .999 .999 .000 1.000 1.000 1.000 1.000 1.000	0.756 .703 .643 .578 .511 .444 .378 .316 .259 .208 .163 .124 .093 .068 .048 .033 .022 .014 .009 .005 .003 .002 .001 .000 .000 .000	-0.244285315332338334299272241208174142086046032014009005001000000000000	0 .031 .067 .108 .154 .206 .262 .323 .388 .454 .522 .588 .652 .711 .765 .813 .855 .890 .918 .940 .958 .940 .958 .995 .995 .997 .995 .997 .999 .999 .999	0.146 .167 .192 .218 .244 .271 .295 .326 .335 .326 .335 .326 .329 .335 .326 .309 .256 .224 .190 .157 .126 .099 .075 .040 .013 .005 .001 .000 .000 .000 .000	0.102 .115 .126 .133 .134 .127 .111 .087 .055 .016 066 104 155 168 160 147 128 108 108 038 052 038 052 012 005 005 005 005 005 005 005 005 005 005 005 005 005 005 006 005 006 005 006 005 006 0

(34)
$$T_{\infty}/T_{W} = 2$$
; Eu = 0; $f_{W} = -1.0$
 $\frac{\delta^{*}\sqrt{Re}}{x} = 4.931$; $\frac{\delta_{1}\sqrt{Re}}{x} = 2.109$; $\frac{\delta_{C}\sqrt{Re}}{x} = 2.167$

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								John S. C.	مر مرکز
η	f	f†	f"	f"¹	θ	θ'	θ"	u∕V∞	ρυ ρ _∞ V _∞
0 .4680246048260482604826112223334445556666778888996420864208642111111111111111111111111111111111111	-1.000 -1.000998995986980962937965865684603164027127644016117391350135013501350135013501355	0 .005 .010 .016 .023 .029 .037 .044 .052 .070 .091 .113 .137 .163 .190 .218 .246 .275 .303 .355 .379 .400 .420 .436 .451 .463 .473 .481 .492 .497 .500 .500 .500	0.024 .026 .028 .030 .033 .035 .037 .040 .042 .048 .053 .062 .066 .069 .071 .069 .066 .062 .057 .051 .045 .039 .033 .027 .022 .017 .010 .005 .000	0.009 .010 .011 .012 .012 .013 .013 .013 .013 .010 .008 .006 .003 006 009 012 015 015 015 015 015 015 015 016 009 001 0010	0 .008 .017 .027 .037 .048 .059 .071 .083 .111 .141 .210 .249 .291 .334 .380 .427 .475 .523 .570 .661 .744 .781 .815 .845 .916 .948 .969 .982 .990 .995 .997 .998 .999 .999 .999 .999 .999	0.041 .043 .046 .049 .052 .055 .058 .062 .072 .079 .087 .094 .101 .112 .116 .119 .120 .119 .120 .119 .103 .080 .089 .080 .072 .063 .047 .032 .021 .007 .004 .000 .000	0.013 .013 .014 .015 .015 .016 .017 .017 .018 .018 .018 .019 .001 .001 .001 .001 .001 .001 .001	0 .005 .011 .017 .023 .031 .039 .047 .057 .078 .103 .132 .166 .203 .245 .291 .340 .392 .446 .502 .665 .715 .761 .803 .840 .968 .998 .998 .998 .998 .998 .998 .998	0 .010 .021 .033 .045 .059 .073 .089 .105 .141 .226 .274 .325 .380 .436 .493 .550 .605 .659 .710 .758 .801 .839 .8702 .926 .946 .946 .983 .995 1.000 1.000

(35)
$$T_{\infty}/T_{W} = 2$$
; Eu = 0.05; $f_{W} = -1.0$
 $\frac{\delta^{*}\sqrt{Re}}{x} = 2.985$; $\frac{\delta_{1}\sqrt{Re}}{x} = 1.908$; $\frac{\delta_{C}\sqrt{Re}}{x} = 2.299$

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η	f	f'	f"	f"t	. θ	θ1	θ 11	u/U∞	ρυ ρ _ω U _∞
0 .4680.482600.482600.482600.482600.482600.482600.482600.482600.482600.482600.482600.482600.482600.482600.482600.482600.482600.482600.482600.4826000.4826000.4826000000000000000000000000000000000000	-1.000998993984971955935911883852816735639529406270124 .033 .198 .372 .551 .736 .925 1.118 1.510 1.708 1.908 2.108 2.308 2.708 2.108 2.308 2.708 2.908 3.108 3.508 3.708 4.108 4.508 4.708 4.908 5.108	0 .018 .036 .054 .073 .091 .110 .129 .148 .166 .185 .222 .258 .292 .324 .353 .380 .403 .424 .4456 .468 .478 .491 .497 .499 .500 .500 .500	0.089 .090 .091 .092 .093 .094 .094 .094 .093 .091 .083 .077 .070 .063 .048 .040 .033 .027 .016 .012 .008 .005 .005 .000 .000	0.005 .005 .005 .004 .004 .003 .002 .000 001 003 016 018 019 019 018 017 016 014 012 010 014 012 010 003 004 001 -	0 .014 .029 .045 .080 .099 .1140 .184 .283 .336 .446 .5558 .659 .749 .7825 .8856 .975 .985 .995 .997 .998 .999 .999 .999 .999 .999 .999	0.069 .073 .078 .082 .087 .092 .097 .102 .106 .111 .115 .123 .130 .135 .138 .139 .138 .134 .129 .122 .114 .094 .094 .094 .094 .094 .045 .037 .030 .023 .018 .011 .008 .001 .000 .000 .000	0.021 .022 .023 .024 .024 .024 .023 .022 .021 .018 .015 .010 .005 011 016 020 023 025 023 025 026 025 023 021 019 017 014 012 009 001 001 001 001 001 001 001 001 001 001 003 003 001 001 001 000 001 001 000 001 000 001 000 001 000	0 .018 .037 .057 .077 .099 .121 .144 .168 .193 .219 .274 .330 .450 .570 .682 .778 .815 .855 .911 .934 .961 .978 .984 .996 .997 .998 .999 .999 .999 .999 .999 .999	0 .036 .072 .109 .146 .183 .220 .258 .295 .330 .370 .444 .515 .583 .647 .706 .759 .848 .883 .912 .935 .970 .989 .998 .998 .998 .998 .998 .998 .99

(36)
$$T_{\infty}/T_{W} = 2$$
; Eu = 0.15; $f_{W} = -1.0$

$$\frac{\delta * \sqrt{Re}}{x} = 1.989; \quad \frac{\delta_1 \sqrt{Re}}{x} = 1.710; \quad \frac{\delta_c \sqrt{Re}}{x} = 2.343$$



		х		24		, x		Jan Wa	مسرم مما
η	f	f¹	f"	f"	θ	θ'	θ"	u/U∞	ρυ Ρωθο
0 111.6802468024680246802482604826048260482604	-1.000997987970948920885800636572504433280199116030058147238280199116030058147238280199116030521617714811910 1.008 1.107 1.207 1.406 1.606 1.806 2.206 2.206 2.206 2.206 3.606 2.206 3.606 3.206 3.406 2.206 3.606 3.206 3.406 3.606 3.206 3.606 3.206 3.606 3.206	0 .033 .066 .097 .128 .157 .186 .239 .263 .286 .398 .348 .366 .397 .410 .423 .453 .460 .4453 .460 .497 .499 .500	0.168 .164 .160 .155 .150 .145 .139 .133 .106 .099 .092 .085 .078 .071 .065 .058 .052 .047 .037 .037 .037 .028 .024 .020 .017 .014 .012 .009 .009 .009 .009 .009 .009 .009 .00	-0.019020024026028035035035035035035035035030029020029021011010008004	0 .018 .038 .038 .080 .103 .128 .153 .208 .236 .236 .296 .327 .358 .3484 .515 .663 .6667 .712 .736 .736 .736 .759 .821 .821 .828 .838 .920 .939 .939 .990 .990 .900 .900 .900 .900 .900	0.089 .094 .100 .107 .113 .125 .131 .136 .141 .146 .149 .153 .155 .157 .158 .158 .156 .154 .151 .148 .144 .140 .129 .124 .118 .111 .105 .099 .092 .086 .074 .062 .051 .008 .001 .000 .000 .000 .000 .000	0.029	0 .034 .1038 .1209 .2452 .3354 .466 .5364 .5364 .5592 .7363 .7366	0 .066 .131 .255 .314 .371 .427 .527 .617 .658 .696 .731 .764 .845 .845 .845 .921 .9347 .947 .947 .947 .947 .947 .947 .947 .9

(37)
$$T_{\infty}/T_{W} = 2$$
; Eu = 0.4; $f_{W} = -1.0$
 $\frac{\delta^{*}\sqrt{Re}}{x} = 1.245$; $\frac{\delta_{1}\sqrt{Re}}{x} = 1.476$; $\frac{\delta_{C}\sqrt{Re}}{x} = 2.320$

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		X		. X		21		NA	مر کریس
η	f	f'	f"	f"'	θ	. 01	θ"	u∕V∞	ρυ _{Ρω} υ _∞
0 .46802468024680260482604 11.1.2.2.2.2.2.3.3.3.3.4.4.4.4.5.5.5.5.6.6.6.7.7.8.8.8.9.9.0.0.1.1.2.2.1.2.2.2.2.2.3.3.3.3.3.3.4.4.4.4.5.5.5.5.6.6.6.7.7.8.8.8.9.9.0.0.1.1.2.2.1.2.2.2.2.2.3.3.3.3.3.3.3.3.3.3	-1.000 994 978 978 916 819 759 693 621 545 464 379 201 108 201 108 201 108 277 277 277 1.177 1.577 1.777 1.577 1.777 1.577 1.777 1.577 2.577 2.577 2.577 2.577 3.577 3.577 3.577 3.577 3.577 3.577 3.577 3.577 3.577 3.777 4.577 4.577	0 .056 .108 .157 .2043 .241 .345 .371 .4459 .4459 .4459 .4494 .4994 .4994 .4998 .500 .500	0.287 .270 .253 .235 .216 .197 .178 .160 .142 .125 .109 .080 .056 .046 .037 .030 .023 .017 .012 .008 .001	-0.079084092094095088083078072065059053047042037024020017014	0 .021 .045 .069 .096 .124 .154 .186 .219 .253 .324 .361 .398 .435 .471 .507 .543 .577 .611 .643 .674 .703 .731 .752 .805 .880 .995 .997 .988 .999 .999 .999 .999 .999 .999	0.103 .111 .120 .129 .138 .146 .154 .162 .168 .174 .183 .184 .183 .184 .182 .179 .175 .170 .164 .158 .151 .127 .111 .094 .051 .040 .031 .023 .017 .012 .000 .000 .000	0.041 .043 .044 .043 .042 .039 .035 .030 .025 .019 .013 .007 .000 012 017 022 027 031 034 034 042 040 042 040 042 040 017 014 017 014 010 006 001 006 001 001 001 000	0 057 113 168 2274 373 4265 549 659 659 6727 777 7795 835 8867 887 999 999 999 999 999 999 999 1.000 1.000 1.000	0 .111 .216 .314 .404 .486 .561 .629 .829 .829 .864 .918 .939 .956 .977 .997 .997 .997 .9997

7

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

(38)
$$T_{\infty}/T_{W} = 2$$
; Eu = 0.5; $f_{W} = -1.0$
 $\frac{\delta \sqrt[4]{Re}}{x} = 1.114$; $\frac{\delta_{i}/\overline{Re}}{x} = 1.428$; $\frac{\delta_{c}/\overline{Re}}{x} = 2.289$

NACA. ρ∞Ü∞ u/U∞ f! θ1 ſ η 0.104 0.046 0.320 -0.100 0 -1.000 .063 .124 .2 -.994 .022 .114 .048 -.108 .062 .300 .125 .240 .049 .124 -.113 .046 .4 .120 .278 -.976 .133 .049 .185 .346 -.117 .071 .173 .255 .6 -.946 -.907 -.858 .048 .243 .443 .222 .231 -.118 .099 .143 .8 .300 .153 .531 .046 .208 -:117 .128 1.0 .265 -.114 .162 .043 .353 .609 .160 -.801 .305 .184 1.2 .193 .405 .678 .162 -.109 .170 .038 .339 1.4 -.736 .454 .739 -.103 .032 -.665 .228 .177 1.6 .369 .141 .791 -.096 .026 .500 .264 .396 .121 .182 -.589 1.8 .543 -.088 .836 -.508 .301 .187 .019 .418 .102 2.0 .584 .012 .873 -.079 .338 .190 -.422 .437 .086 2.2 .192 .004 .622 .904 -.071 .377 -.333 .452 .071 2.4 -.003 .658 .930 .057 -.063 .415 .192 .465 -.241 2.6 .691 .950 -.055 -.010 .453 .191 .045 -.147 .475 2.8 -.022 -.041 .529 .184 .748 .979 .046 .489 .026 3.2 -.030 .994 .497 .012 .600 .173 -.032 .795 .243 3.6 .999 -.039 .833 .500 -.021 .667 .159 4.0 .443 .002 -.043 -.045 .863 .142 .727 .643 4.4 .890 .843 1.043 .781 .125 4.8 -.044 .913 .827 .107 5.2 .866 -.042 .932 .089 5.6 1.243 -.039 .949 1.443 .899 .073 6.0 -.034 .962 .058 .925 6.4 1.643 -.029 .972 .946 .046 6.8 1.843 .980 .962 .035 -.025 7.2 2.043 -.020 .986 .974 .026 2.243 7.6 -.015 .990 .983 .019 2.443 8.0 .989 .013 -.012 .994 8.4 2.643 .993 -.009 .996 .009 2.843 8.8 .006 -.006 .997 3.043 .996 9.2 .998 1.000 .004 -.004 .998 9.6 3.243 .002 -.003 .999 3.443 10.0 -.002 .999 3.643 1.000 .001 10.4 -.001 -.001 1.000 1.000 .001 10.8 3.843 1.000 .000 1.000 11.2 4.043 1.000 .000 .000 1.000 4.243 11.6 1.000 .000 1.000 .000 12.0 4.443

(39)
$$T_{\infty}/T_{W} = 2$$
; Eu = 1; $f_{W} = -1.0$
 $\frac{\delta^*\sqrt{Re}}{x} = 0.793$; $\frac{\delta_{i}\sqrt{Re}}{x} = 1.289$; $\frac{\delta_{c}\sqrt{Re}}{x} = 2.206$

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									, ·
η	f	f¹	f"	f"'	θ	θι	θ"	ս∕Մ∞	ρυ <u>ρω</u> Uω
0 .46802468026048260481.112222223333444555666.48899.00.48	-1.000992967928876813662576485391294196096 .004 .104 .404 .604 .1.204 .604 1.404 1.404 1.604 1.404 1.604 2.204 2.604 2.604 2.604 2.604 3.604 3.604 3.604 4.004	0 .084 .161 .229 .288 .338 .414 .442 .463 .478 .499 .500	0.441 .402 .361 .317 .273 .231 .190 .055 .043 .025 .010 003	-0.181 202 215 217 208 194 176 157 137 117 0083 070 058	0 .022 .046 .074 .104 .137 .173 .211 .251 .293 .336 .425 .513 .555 .596 .673 .741 .800 .850 .945 .945 .963 .976 .985 .999 .999 .999 .999 .999 .999 .999	0.102 115 129 144 158 172 185 197 206 214 219 221 222 219 215 209 135 111 089 052 038 027 018 002 001 000 000	0.063 .068 .071 .073 .072 .068 .062 .053 .043 .031 .019 .007 026 035 043 053 053 059 053	0 .086 .168 .245 .318 .384 .446 .502 .553 .598 .639 .675 .706 .777 .798 .836 .970 .924 .944 .960 .972 .988 .999 .995 .997 .998 .999 1.000 1.000 1.000 1.000 1.000 1.000 1.000	0 .169 .321 .575 .676 .883 .956 .978 .998 .998 1.000

(40) $T_{\infty}/T_{W} = 4$; Eu = 0; $f_{W} = -1.0$

 $\frac{\delta^*/Re}{x} = 6.409; \frac{\delta_1/Re}{x} = 4.161; \frac{\delta_c/Re}{x} = 4.002$

	· х		•		•	ĸ		and NA	CA
η	f	f'	f"	f"1	θ	θ'	θ"	u/V 🗫	ρυ ρωυσ
0	-1.000 -1.000 -1.000 -1.999 -998 -998 -998 -998 -998 -998 -998	0 .003 .005 .008 .011 .018 .021 .028 .032 .036 .040 .049 .058 .067 .077 .086 .106 .125 .144 .152 .161 .199 .214 .219 .223 .224 .239 .243 .248 .249 .250 .250	0.012 .013 .014 .015 .015 .016 .017 .018 .019 .020 .021 .022 .023 .023 .024 .024 .024 .024 .024 .024 .023 .023 .020 .019 .010 .017 .016 .017 .016 .017 .016 .017 .016 .019 .020 .021 .020 .020	0.004 .004 .004 .004 .004 .003 .003 .003	0 .005 .011 .017 .023 .037 .044 .052 .060 .069 .077 .087 .149 .172 .1973 .2250 .277 .306 .3365 .425 .485 .545 .545 .545 .545 .574 .6020 .657 .683 .7316 .815 .850 .8811 .928 .999 .999 .999 .9999 .9999 .9999 .9999 .9999 .9999 .9999 .9999 .9999 .9999 .9999 .9990 .9900 .9000 .900	0.026 .028 .029 .031 .032 .034 .036 .038 .039 .0415 .045 .057 .060 .063 .066 .072 .075 .076 .075 .075 .075 .076 .075 .075 .076 .075 .075 .076 .075 .075 .076 .075 .075 .076 .075 .075 .076 .076 .076 .076 .076 .076 .075 .076 .076 .076 .076 .076 .076 .076 .076	0.007 .008 .008 .008 .008 .009 .009 .009 .009	0 .003 .006 .009 .012 .016 .024 .034 .039 .045 .039 .045 .039 .045 .039 .045 .039 .045 .039 .045 .039 .045 .039 .045 .039 .045 .039 .045 .039 .045 .039 .045 .039 .045 .039 .046 .039 .046 .039 .046 .039 .046 .039 .046 .039 .046 .039 .046 .039 .046 .039 .046 .039 .046 .039 .046 .046 .046 .046 .046 .046 .046 .046	0 .010 .021 .033 .045 .057 .071 .084 .099 .145 .161 .196 .345 .462 .538 .462 .538 .462 .5538 .610 .644 .6778 .765 .791 .858 .876 .894 .995 .999 .999 .999 .999 .999 .999 .9

(41) $T_{\infty}/T_{W} = 4$; Eu = 0.05; $f_{W} = -1.0$ $\frac{\delta * \sqrt{Re}}{x} = 3.405$; $\frac{\delta_{1} \sqrt{Re}}{x} = 3.859$; $\frac{\delta_{C} \sqrt{Re}}{x} = 4.248$

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η	f	f¹	f"	f"	θ	θ;	θ"	u/U _∞	$\frac{\rho u}{\rho_{\infty} U_{\infty}}$
0	-1.000 996 983 963 935 935 900 859 759 638 5700 425 5700 425 096 347 096 347 186 096 186 1	0 .041 .060 .078 .139 .151 .163 .173 .182 .199 .201 .218 .223 .235 .243 .244 .245 .250 .250 .250	0.054 .052 .049 .046 .043 .038 .035 .032 .030 .027 .025 .023 .021 .019 .017 .015 .014 .010 .009 .008 .007 .006 .005 .003 .000	-0.006007007007007007006006006005005005004004004003003003003003001001	0 .021 .044 .070 .097 .125 .187 .220 .253 .322 .358 .322 .358 .322 .358 .322 .358 .322 .358 .322 .358 .323 .483 .497 .5364 .5627 .656 .656 .656 .713 .735 .826 .861 .891 .937 .936 .937 .936 .937 .938 .939 .939 .939 .939 .939 .939 .939	0.051 .056 .061 .070 .074 .077 .080 .083 .085 .087 .088 .088 .088 .088 .087 .085 .083 .081 .079 .076 .073 .069 .066 .062 .055 .048 .028 .023 .014 .010 .000 .000 .000 .000	0.012 .012 .011 .011 .010 .009 .007 .006 .005 .003 .002 .000 001 005 006 007 008 009 009 009 009 009 009 009 009 009 009 005 005 005 005 005 005 005 005 005 009 000 005 005 005 006 007 009 009 009 009 009 009 009 005 005 005 005 005 005 005 005 005 005 009 009 009 009 009 009 009 009 009 000 005	0 .023 .047 .073 .101 .131 .162 .196 .229 .266 .303 .340 .378 .416 .454 .492 .5366 .635 .668 .728 .755 .781 .826 .864 .896 .940 .940 .940 .940 .940 .940 .940 .940	0 .085 .165 .241 .313 .380 .443 .501 .555 .605 .692 .730 .765 .824 .8573 .8911 .927 .941 .953 .963 .994 .994 .994 .9994 .900

(42)
$$T_{\infty}/T_{W} = 4$$
; Eu = 0.15; $f_{W} = -1.0$

$$\frac{\delta^{*}\sqrt{Re}}{x} = 2.040; \frac{\delta_{1}\sqrt{Re}}{x} = 3.640; \frac{\delta_{C}\sqrt{Re}}{x} = 4.309$$

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η	f	f'	f"	f'''	θ	θ1	θ"	u∕U∞	ρυ ρωυ _ω
0 .4680482604826048260420864208642086420 11.1.2223334.455566667788889900.122334.1516687.122334.12234.122234.12234.12234.12224.12234.12224.12	-1.000 998 9983 9953 9954 935 8863 701 6266 374 1997 2891 2990 9900 	0 .039 .056 .072 .082 .127 .138 .148 .1895 .215 .222 .234 .244 .244 .245 .250 .250	0.103 .096 .099 .078 .078 .063 .058 .049 .042 .035 .021 .014 .012 .009 .007 .006 .004 .003 .000	-0.033 032 031 030 028 027 026 023 022 020 015 013 011 010 008 007 006 005 004 003 002 002 002 002	0 .013 .027 .042 .057 .073 .089 .106 .124 .160 .197 .236 .275 .315 .355 .4374 .512 .549 .584 .651 .682 .712 .740 .766 .790 .813 .834 .853 .836 .914 .936 .953 .967 .998 .999 .998 .999 .998 .999 .999 .99	0.066 .069 .072 .075 .078 .080 .083 .086 .089 .099 .095 .098 .099 .100 .100 .098 .096 .094 .091 .087 .084 .080 .076 .072 .067 .063 .059 .054 .050 .046 .038 .031 .024 .019 .014 .011 .008 .006 .004 .003 .000 .000 .000 .000 .000	0.015 .015 .015 .015 .015 .015 .014 .014 .013 .012 .011 .010 .009 .007 .005 .003 .001001006007008009010011011011011011011011011011011011011011010009008007006005006005006005001001001001001000 .000	0 .021 .042 .063 .085 .107 .129 .151 .174 .196 .219 .265 .310 .444 .486 .566 .638 .672 .731 .757 .781 .803 .823 .841 .858 .874 .888 .934 .950 .964 .973 .981 .999 .990 .900 .900 .900 .900 .900 .900 .900 .900 .900 .900 .900 .900 .900 .90	0 .080 .155 .224 .290 .350 .406 .458 .559 .665 .7779 .822 .8859 .914 .9352 .965 .976 .996 .998 .998 .998

> (43) $T_{\infty}/T_{W} = 4$; Eu = 0.4; $f_{W} = -1.0$ $\frac{\delta^{*}\sqrt{Re}}{x} = 1.156$; $\frac{\delta_{1}\sqrt{Re}}{x} = 3.362$; $\frac{\delta_{C}\sqrt{Re}}{x} = 4.266$

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u/U_∞ Palla 01 θ f" f¹ f ή 0.073 0.022 0.170 -0.081 -1.000 0 .034 .130 .154 .022 .015 .077 -.078 .032 .2 -.997 .247 .068 .081 .022 .139 -.075 .031 -.987 .062 .101 .021 .352 .047 .086 -.071 .6 -.972 .088 .124 .133 .446 .065 .090 .020 -.066 .111 .112 -.952 .8 .530 .166 .094 .019 -.062 .083 .098 .132 -.928 1.0 .197 .017 -.057 .103 .097 -.899 .151 .086 1.2 .228 .668 .016 .101 -.052 .122 .167 .075 -.868 1.4 .724 .258 -.047 -.042 -.037 .104 .014 .143 -.833 .181 .066 1.6 .773 .288 .164 .012 .106 .057 -.795 .193 1.8 .011 .317 .815 .185 .109 .204 .049 -.756 2.0 .851 .207 .345 -.034 .111 .009 .213 -.714 .042 2.2 .882 .007 .372 .230 -.030 .112 .220 .035 -.671 2.4 .908 .005 .399 .252 .114 -.026 .227 .030 -.626 2.6 .929 .004 .424 -.023 .114 .232 .275 .025 -.580 2.8 .962 .000 .472 .115 -.018 .321 -.485 .240 .017 3.2 .115 .983 -.003 .517 -.014 .367 .010 -.388 3.6 .246 .996 .557 .413 .113 -.005 -.011 -.289 .005 .249 4.0 1.000 .594 -.007 .002 -.008 .458 .111 -.189 .250 4.4 -.009 .627 .501 .108 -.006 .250 -.002 -.089 4.8 -.011 .658 .544 .104 .011 5.2 .689 -.012 .584 .099 .111 5.6 -.013 .718 .623 .094 .211 6.0 .746 .660 .089 -.013 .311 6.4 .772 -.014 .694 .084 .411 6.8 .727 .078 -.014 .796 7.2 .511 .757 -.014 .818 .072 7.6 .611 -.014 .839 .784 .067 8.0 .711 .810 -.014 .858 .061 .811 8.4 .876 -.013 .833 .056 .911 8.8 -.013 .892 .051 .855 1.011 9.2 .874 -.012 .906 .046 9.6 1.111 .919 .041 -.011 .891 1.211 10.0 .036 .906 -.011 .931 1.311 10.4 .950 .932 .028 -.009 1.511 11.2 -.007 .965 .021 .952 12.0 1.711 .016 -.006 .976 .967 1.911 12.8 -.005 .984 .978 .012 2.111 13.6 -.004 .990 .986 .008 2.311 2.511 14.4 -.003 .994 .991 .005 15.2 -.002 .997 .003 .995 2.711 16.0 -.001 -.001 .997 .999 .002 2.911 16.8 1.000 .998 .001 17.6 3.111 .001 -.001 .999 3.311 18.4 .000 .000 1.000 19.2 3.511 1.000 .000 .000 3.711 20:0 1.000 .000 .000 3.911 20.8 .000 .000 1.000 4.111 21.6

(44)
$$T_{\infty}/T_{W} = 4$$
; Eu = 0.5; $f_{W} = -1.0$ NACA
$$\frac{\delta^{*}\sqrt{Re}}{x} = 1.016; \frac{\delta_{1}\sqrt{Re}}{x} = 3.299; \frac{\delta_{c}\sqrt{Re}}{x} = 4.254$$

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η	f	ft	f"	f"	θ	θ'	θ"	u/U∞	ρυ ρω∪ω
0 .4680244826048260482604 11.222233344.82604882604820864208 11.222233344.85566667788889960420864208 11.222233344.8260482208864208864208	-1.000 986 996 996 996 996 996 896 896 646 554 646 454 154 154 154 146	0 .036 .068 .096 .122 .144 .163 .180 .216 .224 .231 .247 .250 .250	0.188 .170 .152 .135 .118 .103 .089 .077 .065 .046 .038 .031 .020 .011 .004 001	-0.093 091 088 084 079 066 060 054 048 033 025 019 015 011	0 .015 .031 .047 .065 .084 .103 .124 .145 .166 .188 .211 .234 .281 .329 .376 .424 .470 .515 .558 .599 .638 .675 .710 .7472 .779 .824 .847 .868 .886 .903 .917 .941 .959 .999 .999 .999 .999 .999 .999 .99	0.072 .077 .081 .087 .091 .096 .100 .103 .107 .110 .112 .115 .116 .119 .119 .119 .117 .114 .110 .095 .090 .084 .072 .066 .060 .054 .049 .044 .039 .034 .026 .019 .010 .007 .001 .001 .000 .000 .000	0.024 .025 .024 .023 .021 .020 .018 .016 .014 .012 .010 .007 .004 .000003006009011015	0 .037 .074 .110 .146 .180 .214 .247 .278 .309 .338 .367 .394 .445 .491 .533 .569 .604 .638 .670 .701 .758 .883 .871 .888 .903 .917 .930 .940 .959 .982 .989 .994 .997 .999 1.000 1.000 1.000 1.000 1.000	0 .143 .272 .386 .487 .576 .653 .776 .824 .898 .9266 .990 1.000

(45)
$$T_{\infty}/T_{W} = 4$$
; Eu = 1; $f_{W} = -1.0$
 $\frac{\delta^{*}\sqrt{Re}}{x} = 0.715$; $\frac{\delta_{1}\sqrt{Re}}{x} = 3.186$; $\frac{\delta_{C}\sqrt{Re}}{x} = 4.101$

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	,	X		Х -				NAC	
η	f	f'	f"	f"	θ	θι	θ"	u/U∞	ρυ ρωŪ∞
0 .4 .6 .8 .0 .2 .4 .6 .8 .2 .6 .0 .4 .8 .2 .2 .6 .0 .4 .8 .2 .6 .0 .4 .8 .2 .2 .6 .0 .4 .8 .2 .2 .6 .0 .4 .8 .2 .2 .6 .0 .4 .8 .2 .2 .6 .0 .4 .8 .2 .2 .6 .0 .4 .8 .2 .2 .6 .0 .4 .8 .2 .2 .6 .0 .4 .8 .2 .2 .0 .0 .4 .2 .2 .2 .2 .2 .2 .2 .2 .2 .2 .2 .2 .2	2.821 3.021 3.221		0.246 .220 .193 .166 .140 .093 .072 .054 .038 .024 .001 008	-0.127133135133127119086097065056048042	0 .013 .027 .043 .060 .079 .099 .120 .143 .166 .190 .215 .241 .267 .294 .348 .401 .4545 .555 .602 .646 .688 .726 .794 .823 .849 .872 .893 .911 .926 .940 .951 .969 .975 .989 .994 .997 .999 1.000 1.000 1.000 1.000 1.000	0.062 .068 .075 .083 .090 .097 .103 .109 .115 .120 .132 .133 .134 .133 .134 .133 .130 .126 .121 .107 .100 .092 .055 .048 .042 .036 .031 .026 .031 .026 .022 .018 .015 .000 .000 .000	0.034 .035 .036 .036 .035 .034 .031 .029 .026 .022 .019 .015 .000 005 009 015 017 018 019 019 019 019 019 019 016 015 016 017 018 019 019 019 019 019 019 010 016 015 016 016 017 016 016 016 017 016 017 018 019 019 019 019 019 019 019 010 000		0 .187 .352 .496 .619 .722 .805 .871 .958 .982 .996 1.000

(46) $T_{\infty}/T_{W} = 1/2$; Eu = -0.060; $f_{W} = 0$

 $\frac{\delta^* \sqrt{Re}}{x} = 3.043; \frac{\delta_1 \sqrt{Re}}{x} = 0.441; \frac{\delta_c \sqrt{Re}}{x} = 0.348$

η	f	f١	f#	f"'	θ	0'	θ"	u/U _{co}	ρυ ρω ^U ω
0	0	0	0	0.121	0	0.206	0.018	0	0
.1	.000	.001	.012	.126	.021	.208	.019	.001	•000
.2	.000	.002	.025	.132	.042	.210	.019	.002	.001
.3	.000	.006	.039	.139	.063	.212	. 020	.006	.003
.4	.001	.010	.053	.146	.084	.214	.020	.010	.005
.5	.003	.016	.068	.153	.106	.216	.021	.015	.008
.6	.005	.024	.084	.161	.128	.218	.021	.022	.012
.7	.007	.033	.100	.170	.150	.220	.022	.031	.017
.8	.011	.044	.118	.178	.172	.223	.022	.040	.022
.9	.016	.057	.136	.188	.194	.225	.023	.051	.028
.9 1.0	.023	.071	.155	.198	.217	.227	.023	.064	.036
1.1	.030	.088	.176	.209	.240	.229	.023	.077	.044
1.2	.040	.106	.197	.220	.263	.231	.023	.092	.053
1.3	.052	.127	.219	.231	.286	.234	.023	.109	.064
1.4	.066	.150	.243	.243	.309	.236	.022	.127	.075
1.5	.082	.176	.268	.255	.333	.238	.022	.146	.088
1.6	.101	.204	.294	.267	.357	.240	.021	.167	.102
1.7	.123	.235	.321	.280	.381	.244	.020	.190	.117
1.8	.148	.268	.350	.292	.406	.246	.018	.214	.134
1.9	.177	.305	.330	.303	.431	.248	.016	.239	.152
2.0	.209	.344	.410	.313	.456	.250	.014	.266	.172
2.1	.246	.387	.442	.321	.481	.251	.011	.294	.193
2.2	.286	.433	.475	.327	.506	.252	.007	.323	.216
2.3	.332	.482	.508	.330	.531	.252	.003	.354	.241
2.4	.383	.534	.541	.330	.556	.252	002	.3 86	.267
2.5	.439	.590	.573	.326	.581	.252	007	.418	.295
2.6	.501	.649	.606	.315	.607	.251	014	.452	.324
2.7	.569	.711	.636	.299	.632	.249	021	.486	.355
2.8	.643	.776	.665	.275	.656	.247	029	.521	.388
2.9	.724	.844	.691	.243	.681	.243	047	.592	.457
3.0	.812	.914	.713	.203	.705 .729	.239	057	.627	.493
3.1	.907	.986	.731		.752	.228	068	.662	.530
3.2	1.010	1.060	.744	.095		.220	078	.696	.567
3.3	1.119	1.135	.750	.029	.774	.212	089	.728	.605
3.4	1.236	1.210	.749	044	.817	.202	099	.760	.642
3.5	1.361	1.284	.741	202	.836	.192	108	.790	.679
3.6	1.493	1.358 1.429	.701	280	.855	.181	116	.818	.715
3.7	1.779	1.498	.669	354	.872	.169	123	.844	.749
3.8	1.932	1.563	.630	419	.889	.156	128	.868	.781
4.0	2.092	1.624	.586	472	.904	.143	131	.890	.812
4.1	2.257	1.680	.536	511	.917	.130	131	.909	.840
4.2	2.427	1.731	.484	534	.930	.117	130	.926	.865
4.3	2.603	1.776	.430	540	.941	.104	127	.941	.888
4.4	2.782	1.817	.376	530	.950	.092	121	.953	.908
4.5	2.966	1.852	.324	506	.959	.080	115	964	.926
4.6	3.153	1.882	.275	471	.966	.069	106	:972	.941
4.7	3.342	1.907	.230	428	.973	.059	097	.979	.954
4.8	3.534	1.928	.190	381	.978	.049	087	.985	.964
4.9	3.728	1.945	.154	331	.983	.041	078	989	.973
5.0	3.923	1.959	.124	283	.987	.034	068	.993	.980
5.1	4.119	1.970	.098	236	.990	.028	058	.995	.985
5.2	4.317	1.979	.076	194	.992	.022	049	.997	.989
5.3	4.515	1.985	.059	194 157	.994	.018	041	.998	.993
5.4	4.714	1.990	.045	125	.996	.014	034	.999	.995
5.5	4.913	1.994	.034	098	.997	.011	027	1.000	.997
5.6	5.113	1.997	.025	075	.998	.008	022	1.001	.999
5.7	5.312	1.999	.018	057	.999	.006	018	1.001	1.000
5.8	5.512	2.001	.013	044	.999	.005	014	1.001	1.000
5.9	5.713	2.002	.010	032	1.000	.004	011	1.001	1.001
6.0	5.913	2.003	.007	024	1.000	.003	008	1.001	1.002
6.1	6.113	2.004	005	018	1.000	.002	006	1.001	1.002
6.2	6.314	2.004	.003	013	1.001	.001	004	1.001	1.002
6.3	6.514	2.004	.002	009	1.001	.001	003	1.001	1.002
6.4	6.714	2.004	.001	006	1.001	.001	002	1.001	1.002
		2.004	.001	004	1.001	.000	001	1.001	1.002
6.5	6.915 7.115	2.005	.000	003	1.001	.000	001	1.001	1.002

TABLE I - VELOCITY, WEIGHT-FLOW, AND TEMPERATURE DISTRIBUTIONS IN LAMINAR BOUNDARY LAYER WITH VARIABLE FLUID PROPERTIES, PRESSURE GRADIENT IN MAIN STREAM, AND FLOW THROUGH POROUS WALL - Continued

THROUGH POROUS WALL - Continued

NACA

(47) $T_{\infty}/T_{W} = 1/2$; Eu = -0.04; $f_{W} = 0$

 $\frac{\delta^* \sqrt{Re}}{v}$ = 2.309; $\frac{\delta_1 \sqrt{Re}}{v}$ = 0.406; $\frac{\delta_c \sqrt{Re}}{v}$ = 0.420

(48)
$$T_{\infty}/T_{W} = 1/2$$
; Eu = 0; $f_{W} = 0$

$$\frac{\delta^{*}/Re}{x} = 1.898$$
; $\frac{\delta_{1}/Re}{x} = 0.347$; $\frac{\delta_{C}/Re}{x} = 0.457$

									ρu
η	f	f1	f"	f"	θ	81	θ"	u/U∞	$\rho_{\infty}U_{\infty}$
0	0	0	0.346	0.136	0	0.290	0.036	0	0
.1	.002	.035	.360	.145	.029	.294	.037	.035	.018
.2	.007	.072	.375	.154	.059	.297	.038	.070	.036
•3	.016	.110	.391	.164	.089	.301	.039	.106	.055
.4	.029	.150	.408	.173	.119	.305	.039	.141	.075
.5	.046	.192	.426	.183	.150	.309	.039	.178	.096
.6 .7	.068	.236 .281	.444	.192	.181	.313 .317	.037	.251	.140
.8	.124	.328	.484	.208	.244	.320	.034	.288	.164
.9	.159	.378	.506	.216	.276	.324	.032	.326	.189
1.0	.200	.429	.527	.221	.309	.327	.028	.363	.215
1.1	.245	.483	.550	.225	.342	.329	.024	.401	.242
1.2	.296	.539	.572	.226	.375	.331	.018	.438	.270
1.3	.353	.598	.595 .617	.225	.408	.333	.011	.476 .513	.299
1.5	.485	.721	.639	.210	.475	.334	006	.550	.361
1.6	.560	.786	.659	.195	.508	.333	016	.586	.393
1.7	.642	.853	.677	.174	.541	.331	028	.622	.426
1.8	.731	.922	.694	.146	.574	.327	041	.657	.461
1.9	.826	.992	.706	.111	.607	.322	055	.691	.496
2.0	.929 1.039	1.063	.715 .720	.065	.639	.316 .308	070 086	.723	.531
2.2	1.156	1.206	.719	038	.701	.299	102	.784	.567 .603
2.3	1.280	1.278	.712	101	.730	.288	119	.812	.639
2.4	1.411	1.349	.698	168	.758	.275	134	.837	.674
2.5	1.550	1.418	.678	236	.785	.261	149	.861	.709
2.6	1.695	1.484	.651	304	.810	.246	162	.883	.742
2.7	1.846	1.548	.618 .578	367 422	.834 .856	.229 .211	173 181	.902	.774 .804
2.9	2.168	1.663	.534	466	.876	.193	186	.934	.831
3.0	2.337	1.714	.486	482	.895	.174	187	.947	.857
3.1	2.510	1.760	.436	512	.911	.155	185	.958	.880
3.2	2.688	1.801	.385	510	.926	.137	180	.967	.901
3.4	3.056	1.837 1.868	.334 .286	496 468	.939	.120 .103	171 160	.975 .981	.919 .934
3.5	3.244	1.894	.241	431	.959	.088	147	.986	.947
3.6	3.435	1.916	.200	387	.967	.074	132	.989	.958
3.7	3.627	1.935	.164	340	.974	.061	118	.992	.967
3.8	3.821	1.949	.132	292	.980	.050	103	.994	.975
3.9 4.0	4.017	1.961	.105 .083	247	.984	.041 .033	088 074	.996 .997	.981 .985
4.1	4.411	1.978	.064	167	.991	.026	062	.998	.989
4.2	4.609	1.984	.049	133	.993	.020	051	.998	.992
4.3	4.808	1.988	.037	105	.995	.016	041	.999	.994
4.4	5.007	1.991	.028	082	.996	.012	033	.999	.996
4.5	5.206 5.405	1.994 1.995	.021	063 048	.997	.009	026	.999	.997
4.7	5.605	1.997	.011	036	.999	.007	020 015	.999 1.000	.998 .998
4.8	5.805	1.998	.008	027	.999	.004	012	1.000	.999
4.9	6.004	1.998	.006	020	1.000	•003	009	1.000	.999
5.0	6.204	1.999	.004	014	1.000	.002	006	1.000	.999
5.1	6.404	1.999	.003	010	1.000	.001	004	1.000	
5.2 5.3	6.604	2.000	.002	008 005	1.000	.001 .001	003 002	1.000	1.000
5.4	7.004	2.000	.001	004	1.000	.000	002	1.000	1.000
5.5	7.204	2.000	.001	003	1.000	.000	001	1.000	1.000
5.6	7.404	2.000	.000	002	1.000	.000	001	1.000	1.000
5.7	7.604	2.000	.000	001	1.000	.000	001	1.000	1.000
5.8 5.9	7.804 8.004	2.000	.000	.000	1.000	.000	001	1.000	1.000
6.0	8.204	2.000	.000	.000	1.000	.000	001 001	1.000	1.000
6.1	8.404	2.000	.000	.000	1.000	.000	001	1.000	1.000

(49)
$$T_{\infty}/T_{W} = 1/2$$
; Eu = 0.5; $f_{W} = 0$

$$\frac{\delta^{*}\sqrt{Re}}{x} = 0.980$$
; $\frac{\delta_{1}/\overline{Re}}{x} = 0.116$; $\frac{\delta_{C}/\overline{Re}}{x} = 0.460$

(50) $T_{\infty}/T_{W} = 1/2$; Eu = 1.0; $f_{W} = 0$

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 $\frac{\delta^* \sqrt{Re}}{x} = 0.768; \frac{\delta_1 \sqrt{Re}}{x} = 0.065; \frac{\delta_c \sqrt{Re}}{x} = 0.415$

		х	0.700,	X	0.000, -	Х			
η	f	f١	f"	fui	θ	θ 1	θ"	u/U _®	ρυ ρ _ω υ _∞
0	0	0	1.800	-0.713	0	0.530	0.119	0	0
.1	.009	.176	1.725	779	.054	.542	.125	.172	.088
.2	,035	.345	1.645	827	.108	.555	.124	.326	.172
.3	.078	.505	1.560	860	.164	.567	.116	.464	.253
.4	.136	.657	1.473	879	.222	.577	.100	.584	.328
.5	.209	.800	1.384	923	.280	.586	.075	.688	.400
.6	.296	.934	1.292	889	.33 9	.592	.041	.775	.467
.7	.3 95	1.058	1.204	881	.3 98	.594	003	.848	.529
.8	.507	1.174	1.116	870	.458	.591	056	.906	.587
.9	.630	1.282	1.030	855	.516	.583	118	.951	.641
1.0	.763	1.380	.945	845	.574	-568	187	.984	.690
1.1	.906	1.471	.862	827	.630	.545	259	1.008	.73 5
1.2	1.057	1.553	.780	815	.683	.516	~.331	1.022	.776
1.3	1.216	1.627	.699	801	.733	.479	3 97	1.031	.813
1.4	1.382	1.693	.619	786	.779	.437	452	1.034	.846
1.5	1.554	1.751	.542	766	.820	.389	491	1.033	.875
1.6	1.732	1.801	.466	742	.856	.339	509	1.030	.900
1.7	1.914	1.844	.394	702	.888	.288	505	1.025	.922
1.8	2.100	1.880	.326	650	.914	.239	481	1.021	.940
1.9	2,290	1.909	.264	588	.936	.193	439	1.016	.955
2.0	2.482	1.933	.209	513	.95 3	.152	3 85	1.012	.966
2.1	2.676	1.951	.161	436	.966	.116	325	1.008	.976
2.2	2.872	1.966	.122	3 59	.976	.087	265	1.006	.983
2.3	3.069	1.976	.090	281	.984	.063	208	1.004	.988
2.4	3.267	1.984	.065	223	.989	.045	159	1.003	.992
2.5	3.466	1.989	.045	164	.993	.031	118	1.002	.995
2.6	3.665	1.993	.031	119	.996	.021	085	1.001	.996
2.7	3.865	1.996	.021	087	.997	.014	059	1.000	.998
2.8	4.064	1.997	.014	060	.999	.009	040	1.000	.999
2.9	4.264	1.998	.009	045	.999	.006	026	1.000	.999
3.0	4.464	1.999	.005	026	1.000	.003	017	1.000	1.000
3.1	4.664	2.000	.003	017	1.000	.002	010	1.000	1.000
3.2	4.864	2.000	.002	010	1.000	.001	006	1.000	1.000
3.3	5.064	2.000	.001	007	1.000	.001	003	1.000	1.000
3.4 3.5	5.264	2.000	.000	003 002	1.000	.000	002	1.000	1.000
3.5	5.464	2.000	.000	002	1.000	.000	001	1.000	1.000

(51)
$$T_{\infty}/T_{W} = 1/4$$
; Eu = 0; $f_{W} = 0$ NACA
$$\frac{\delta^{*}\sqrt{Re}}{x} = 2.031; \frac{\delta_{1}\sqrt{Re}}{x} = 0.182; \frac{\delta_{C}\sqrt{Re}}{x} = 0.246$$

				٠,					
η	f	f'	f"	f"	θ	θ'.	θ"	u/U _∞	<u>ρυ</u> ρ _∞ U _∞
0 .100 .200 .300 .400 .500 .600 .700 .800 .900 1.100 1.200 1.300 1.400 1.500 1.600 1.700 1.800 2.200 2.200 2.200 2.300 2.400 2.500 2.600 2.750 2.850 2.950 3.050 3.055 3.125 3.125 3.175 3.125 3.175 3.125 3.175 3.125 3.175 3.125 3.175 3.125 3.175 3.125 3.175 3.125 3.175 3.125 3.175 3.125 3.175 3.125 3.175 3.125	0 .002 .007 .017 .031 .049 .073 .136 .176 .224 .278 .340 .411 .583 .7924 1.067 1.225 1.402 1.597 1.225 1.402 1.597 1.214 .214 .309 4.499 4.595 4.692	0 .037 .076 .117 .161 .209 .260 .314 .437 .507 .582 .664 .753 .851 .958 1.204 1.346 1.502 1.672 1.857 2.270 2.493 2.722 2.950 3.663 3.270 2.493 2.722 2.950 3.663 3.736	0.356 .378 .402 .428 .458 .491 .528 .568 .614 .665 .721 .784 .854 .934 1.022 1.121 1.352 1.484 1.626 1.775 1.926 2.069 2.273 2.240 2.179 2.240 2.179 2.240 2.179 1.565 1.655 1.080 1	0.208 .254 .282 .3147 .3486 .479 .536 .599 .657 .749 .8387 1.1269 1.269 1.372 1.464 1.569 1.569 1	0 .029 .089 .120 .184 .216 .2284 .315 .2284 .315 .329 .426 .460 .284 .355 .29 .460 .735 .3929 .460 .735 .814 .870 .926 .936 .945 .996 .997 .998 .998 .998 .998 .998 .998 .998	0.288 .294 .300 .306 .319 .325 .340 .347 .354 .361 .386 .390 .392 .390 .385 .376 .392 .390 .256 .238 .219 .180 .161 .142 .124 .107 .092 .084 .071 .065 .059 .054	0.053 .056 .059 .0657 .0672 .0772 .0772 .0772 .0772 .0772 .0755 .0045 .0045 .0055 .0045 .0055 .00545 .0071 .0072 .0072 .0072 .0072 .0073 .0072 .0073 .0073 .0073 .0073 .0073 .0074	0 .036 .072 .109 .147 .185 .224 .263 .304 .344 .385 .429 .511 .5595 .678 .720 .757 .794 .829 .915 .935 .935 .935 .935 .935 .935 .936 .996 .996 .996 .996	0 .009 .019 .0240 .0055 .0079 .1097 .1456 .1813 .2469 .12469 .13375 .4184 .5147 .6281 .7362 .7362 .8464

(51) $T_{\infty}/T_{W} = 1/4$; Eu = 0; $f_{W} = 0$

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 $\frac{\delta^*\sqrt{\text{Re}}}{x} = 2.031; \frac{\delta_1\sqrt{\text{Re}}}{x} = 0.182; \frac{\delta_0\sqrt{\text{Re}}}{x} = 0.246 - \text{Concluded}$

	Х.		. A		X				
η	f	f'	f"	ſ"	θ	θ 1	θ"	u/U _∞	$\frac{\rho u}{\rho_{\infty} U_{\infty}}$
3.225 3.250 3.275 3.300 3.325 3.350 3.350 3.425 3.450 3.455 3.550 3.550 3.575 3.625 3.625 3.625 3.750 3.725 3.750 3.725 3.750 3.725 3.750 3.825 3.825 3.875 3.825 3.875 3.825 3.825 3.875 3.825 3.875 3.825 3.825 3.875 3.825 3.875 3.825	4.788 4.886 4.983 5.081 5.179 5.277 5.376 5.475 5.574 5.673 5.772 5.871 5.971 6.070 6.270 6.470 6.569 6.669 6.769 6.869 6.769 6.869 6.769 7.069 7.170 7.270 7.470 7.570 7.470 7.570 7.870 8.071 8.271 8.471 8.672	3.880 3.895 3.908 3.919 3.930 3.939 3.947 3.955 3.961 3.967 2.3.977 3.985 3.99	0.596 .540 .488 .440 .395 .354 .316 .281 .250 .222 .196 .172 .133 .116 .102 .088 .077 .067 .058 .050 .043 .037 .027 .023 .020 .017 .014 .010 .008 .006 .001 .006 .006 .006 .006 .006 .006	-2.297 -2.154 -2.007 -1.861 -1.717 -1.581 -1.446 -1.308 -1.198 -1.090 979 877 785 702 632 558 492 439 382 439 255 193 217 108 193 170 149 127 108 069 064 044 033 019 013	0.991 .992 .993 .994 .995 .996 .997 .997 .998 .999 .999 .999 .999 .999	0.049 .044 .040 .036 .032 .029 .026 .023 .020 .018 .016 .014 .012 .011 .009 .008 .007 .006 .005 .005 .003 .003 .003 .002 .002 .001 .001 .001	-0.194180166153141129118107097079071064057051045040035031027024020018016014011010009008007004001009008007001002	0.997 .997 .997 .998 .998 .998 .998 .998	0.970 .974 .977 .980 .982 .985 .987 .989 .990 .992 .993 .994 .995 .996 .997 .998 .999 .999 1.000 1.000 1.000 1.000 1.001 1.001 1.001 1.001 1.001 1.002 1.002 1.002 1.002 1.002
4.250 4.300 4.350 4.400 4.450 4.500	8.872 9.072 9.273 9.473 9.674 9.874	4.007 4.007 4.007 4.007 4.007 4.007 4.007	.001 .001 .000 .000 .000	007 013 005 004 001 001	1.001 1.001 1.001 1.001 1.001 1.001	.000 .000 .000 .000 .000	002 003 000 000 000 000	1.000 1.000 1.000 1.000 1.000 1.000 1.000	1.002 1.002 1.002 1.002 1.002 1.002

(52) $T_{\infty}/T_{W} = 1/4$; Eu = 0.5; $f_{W} = 0$

NACA

 $\frac{\delta^* \sqrt{Re}}{x} = 1.033; \frac{\delta_1 \sqrt{Re}}{x} = -0.030; \frac{\delta_c \sqrt{Re}}{x} = 0.271$

		x	1,033;	х	0.030;	x			
η	f	f'	f"	f"	θ	θ'	θ"	u/Մ ₀₀	ρυ ρω ^υ ω
0	0	0	1,930	-0.124	0	0.480	0.147	0	0
.100	.010	.192	1.917	125	.049	.495	.160	.185	.048
•200	.038	.384	1.905	116	.099	.512	.170	.355	. 096
.300	.086	.574	1.895	096	.151	.529	.175	.508	.143
.400	.153	.762	1.886	064	.205	.547	.176	-645	.191
•500	.239	.951	1.882	017	.261	.564	.171	.765	.238
.600	.343	1.139	1.884	.047	.318	.581	.157 .133	.868 .953	.285 .332
.700	.467	1.328	1.893	.137 .235	.377 .437	.595 .607	.09.7	1.021	.380
.800	.609	1.518 1.710	1.911	.355	.498	.614	.046	1.072	.428
.900 1.000	.951	1.906	1.982	.482	.560	.615	024	1.106	.477
1.100	1.152	2.107	2.037	.604	.621	.609	114	1,126	.527
1.200	1.373	2.314	2.101	.671	.681	.592	227	1.132	•578
1.250	1.491	2.419	2.135	.674	.710	.579	291	1.131	.605
1.300	1.615	2.527	2.168	.627	.739	-563	361	1.127	.632
1.350	1.744	2.636	2.197	.522	.766	.543	433 506	1.121	.659 .687
1.400	1.878	2.746 2.858	2.219	.338 .076	.793 .818	.519 .492	577	1.104	.714
1.450 1.500	2.018	2.969	2.224	282	.842	-462	644	1.093	.742
1.550	2.316	3.079	2.199	756	.864	.428	705	1.083	.770
1.600	2.472	3.188	2.148	-1.288	.885	.391	751	1.072	.797
1.650	2.634	3.293	2.069	-1.909	.903	353	784	1.062	.823
1.700	2.802	3.394	1.958	-2.515	.920	.313	796	1.052	.848
1.750	2.974	3.488	1.818	-3.087	.935	.274	789 760	1.043	.872 .894
1.800	3.151	3.574	1.651	-3.522 -3.800	.947 .958	.235 .198	712	1.027	.913
1.850 1.900	3.331	3.652 3.720	1.468	-3.857	.967	.164	647	1.022	.930
1.950	3.704	3.779	1.084	-3.766	.975	.133	575	1.016	.945
2.000	3.894	3.828	.901	-3.554	.981	.107	496	1.012	.957
2.025	3.990	3.849	.814	-3.352	.983	.095	457	1.011	.962
2.050	4.087	3.869	733	-3.183	.985	.084	419	1.009	.967
2.075	4.184	3.886	.655	-3.052	.987	.074	381	1.008	.972 .975
2.100	4.281	3.901	.582	-2.769	.989	.065 .056	344 310	1.007	979
2.125	4.379	3.915 3.927	.516	-2.547 -2.375	.991 .992	.049	278	1.004	982
2.150 2.175	4.477	3.937	.398	-2.109	.993	.043	247	1.004	.984
2.200	4.674	3.946	.348	-1.949	.994	.037	219	1.003	.987
2.225	4.773	3.954	.301	-1.701	.995	.032	193	1.003	.989
2.250	4.872	3.961	.262	-1.508	.996	.027	170	1.002	.990
2.275	4.972	3.967	.226	-1.327	.997	.023	148	1.002	.992 .993
2.300	5.071	3.972	.195	-1.164	.997	.020	129	1.001	.994
2.325	5.171	3.976 3.980	.168	-1.007 883	.998 .998	.017	096	1.001	.995
2.350 2.375	5.270 5.370	3.983	.123	796	.998	.012	082	1.001	.996
2.400	5.470	3.986	.104	716	.999	.010	070	1.000	.996
2,425	5.570	3.988	.088	592	.999	.008	060	1.000	.997
2.450	5.669	3.990	.074	515	.999	.007	051	1.000	.998 .998
2.475	5.769	3.992	.062	426 355	.999	.006	043 036	1.000	•998
2.500	5.869	3.993 3.994	.053	304	1.000	.004	030	1.000	.999
2.525	5.969 6.069	3.995	.038	250	1.000	.003	025	1.000	.999
2.550	6.169	3.996	.032	205	1.000	.003	021	1.000	.999
2.600	6.269	3.996	.027	168	1.000	.002	017	1.000	.999
2.625	6.369	3.997	.024	138 134	1.000	.002	014	1.000	.999
2.650	6.469	3.997	.020	134	1.000	.001	012	1.000	.999
2.675	6.569	3.998	.016	084	1.000	.001	010	1.000	.999
2.700	6.669	3.998	.013	070	1.000	.001	006	1.000	1.000
2.725	6.769	3.998 3.998	.011	046	1.000	.001	005	1.000	1.000
2.750	6.969	3.998	.008	038	1.000	.000.	004	1.000	1.000
2.800	7.069	3.998	.007	029	1.000	.000	003	1.000	1.000
2.825	7.169	3.999	.006	024	1.000	.000	003	1.000	1.000
2.850	7.269	3.999	.005	020	1.000	.000	002	1.000	1.000
2,900	7.469	3.999	.003	011	1.000	•000	001	1.000	1.000
2.950	7.669	3.999	.002	007	1.000	.000	001	1.000	1.000
3.000	7.869	4.000	.002	004	1.000	.000	.000	1.000	1.000
3.050	8.069	4.000	.002	.000	1.000	.000	.000	1.000	1.000
3.100	8.269	±.000		.000	2.000			1 - 1 - 1 - 1	

(53) $T_w/T_w = 1/4$; Eu = 1.0; $f_w = 0$ $\frac{\delta^* \sqrt{Re}}{x} = 0.820$; $\frac{\delta_1 \sqrt{Re}}{x} = -0.064$; $\frac{\delta_C \sqrt{Re}}{x} = 0.246$

.2000	0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0 0
1.000	662
2000	92 .135 89 .200 52 .262 82 .323 7382 46 .440 86 .497 00 .554 612 71 .672 55 .702 21 .763 03 .822 270 .850
3000	89
. \$\frac{400}{.5000} \ .214 \ 1.050 \ 2.461 \759 \ .251 \ .679 \ .245 \ .8 \ .8 \ .6000 \ .472 \ 1.528 \ 2.390 \6467 \ .392 \ .702 \ .218 \ .9 \ .8 \ .8 \ .8 \ .9 \ .9 \ .8 \ .8	52 .262 82 .323 779 .382 46 .440 86 .497 700 .554 93 .612 71 .672 538 .732 21 .763 03 .793 .822 70 .850
.5000	82 .323 .382 .440 .66 .497 .554 .612 .771 .672 .555 .702 .553 .703 .703 .703 .703 .703 .822 .703 .822 .822 .838
.7000	46 .440 86 .497 00 .554 93 .612 71 .672 .702 .702 .703 .793 .822 .703 .825 .850
.8000	86
1.0000	00 .554 93 .612 71 .672 55 .702 38 .732 21 .763 303 .793 86 .822 70 .850
1.0000 1.268 2.450 2.347 .515 .684 .705 356 1.1 1.1500 1.562 2.6687 2.400 .498 .752 .659 582 1.1 1.2000 1.806 2.929 2.429 .000 .815 .588 819 1.1 1.2000 1.955 3.050 2.417 528 .843 .545 930 1.1 1.3500 2.272 3.287 2.294 -2.020 .893 .443 -1.020 1.1 1.4500 2.439 3.399 2.171 -2.936 .914 .388 -1.116 1.0 1.5000 2.789 3.598 1.797 -4.430 .947 .278 -1.058 1.0 1.6250 3.252 3.785 1.322 -4.939 .970 .181 -865 1.0 1.6500 3.346 3.814 1.079 -4.693 .978 .140 -739 1.0 1.6750	93
1.1000 1.525 2.687 2.400 .498 .752 .659 582 1.1 1.2000 1.8662 2.808 2.421 .313 .785 .626 703 1.1 1.2500 1.955 3.050 2.417 528 .843 .545 930 1.1 1.3500 2.110 3.170 2.574 -1.204 .869 .496 -1.020 1.1 1.3500 2.272 3.287 2.294 -2.020 .893 .443 -1.084 1.0 1.4000 2.439 3.599 2.171 -2.936 .914 .588 -1.116 1.0 1.5000 2.789 3.598 1.797 -4.430 .947 .278 -1.058 1.0 1.5500 2.971 3.682 1.565 -4.776 .959 .227 974 1.0 1.6250 3.252 3.785 1.199 -4.855 .974 .160 -804 1.0 1.7250 3.442 3.839 .965 -4.461 .981 .123 -675	71
1.1500	55 .702 38 .732 21 .763 03 .793 86 .822 70 .850
1.2500	21 .763 03 .793 86 .822 70 .850
1.3000 2.110 3.170 2.374 -1.204 .869 .496 -1.020 1.1	03 .793 86 .822 70 .850
1.3500 2.272 3.287 2.294 -2.020 .893 .443 -1.084 1.0 1.4500 2.439 3.399 2.171 -2.936 .914 .388 -1.116 1.0 1.4500 2.612 3.503 2.002 -3.769 .932 .332 -1.06 1.0 1.5500 2.971 3.682 1.565 -4.776 .959 .227 974 1.0 1.6000 3.157 3.754 1.322 -4.939 .970 .181 -865 1.0 1.6550 3.252 3.785 1.199 -4.855 .974 .160 -804 1.0 1.6750 3.442 3.839 .965 -4.461 .981 .123 -675 1.0 1.6750 3.442 3.839 .965 -4.461 .981 .10 -739 1.0 1.7250 3.635 3.882 .759 -3.862 .986 .092 -549 1.0 1.7750 3.830 3.915 .576 -3.278 .990 .068 -4.33 1	86 .822 70 .850
1.4000 2.439 3.599 2.171 -2.936 .914 .388 -1.116 1.00 1.4500 2.612 3.503 2.002 -3.769 .932 .332 -1.106 1.0 1.5000 2.971 3.682 1.585 -4.776 .959 .227 974 1.0 1.6000 3.157 3.754 1.322 -4.939 .970 1.81 865 1.0 1.6250 3.252 3.785 1.199 -4.653 .974 .160 -804 1.0 1.6750 3.442 3.839 .965 -4.461 .981 .123 -675 1.0 1.7250 3.635 3.882 .858 -4.081 .984 .107 -611 1.0 1.7750 3.733 3.900 .664 -3.697 .986 .092 -549 1.0 1.8000 3.928 3.928 .499 -2.957 .992 .057 -379 1.0 1.8250 4.027 3.940 .429 -2.680 .993 .049 -2.32	70 .850
1.4500 2.612 3.503 2.002 -3.769 .932 .332 -1.106 1.0 1.5000 2.789 3.598 1.797 -4.430 .947 .278 -1.058 1.0 1.5500 2.971 3.682 1.565 -4.776 .959 .227 974 1.0 1.6250 3.157 3.781 1.322 -4.939 .970 .181 865 1.0 1.6500 3.342 3.839 .965 -4.855 .974 .160 804 1.0 1.6750 3.442 3.839 .965 -4.461 .981 .123 675 1.0 1.7000 3.538 3.862 .858 -4.081 .984 .107 611 1.0 1.7500 3.733 3.900 .664 -3.697 .986 .092 .559 1.0 1.8750 3.733 3.900 .664 -3.278 .990 .068 -433 1.0 1.8750 4.021 3.950 .366 -2.957 .988 .079 -379	
1.5000 2.789 3.598 1.797 -4.430 .947 .278 -1.058 1.0 1.5500 2.971 3.682 1.565 -4.776 .959 .227 974 1.0 1.6000 3.157 3.754 1.322 -4.939 .970 .181 865 1.0 1.6500 3.346 3.814 1.079 -4.693 .974 .160 804 1.0 1.6750 3.442 3.839 .965 -4.461 .981 .123 675 1.0 1.7000 3.538 3.862 .858 -4.081 .984 .107 611 1.0 1.7250 3.635 3.882 .759 -3.862 .986 .092 -549 1.0 1.7750 3.635 3.992 .499 -2.957 .998 .097 -489 1.0 1.8250 4.027 3.940 .429 -2.957 .992 .057 -379 1.0 1.8500 4.126 3.950 .366 -2.305 .994 .041 -2.26 1	
1.6000 3.157 3.754 1.322 -4.939 .970 .181 865 1.00 1.6500 3.252 3.785 1.199 -4.855 .974 .160 -804 1.0 1.6500 3.344 3.839 .965 -4.461 .981 .123 -675 1.0 1.7000 3.538 3.862 .958 -4.081 .984 .107 -611 1.0 1.7500 3.635 3.862 .558 -4.081 .986 .092 -549 1.0 1.7500 3.635 3.915 .576 -3.278 .990 .068 -433 1.0 1.8000 3.928 .499 -2.957 .992 .057 -379 1.0 1.8250 4.027 3.940 .429 -2.680 .993 .049 -332 1.0 1.8650 4.126 3.950 .366 -2.305 .994 .041 286 1.0 1.8750 4.224 3.958 .358 -2.275 .995 .037 -226 1.0	
1.6250 3.252 3.785 1.199 -4.855 .974 .160 -804 1.0 1.6750 3.546 3.814 1.079 -4.693 .978 .140 739 1.0 1.6750 3.442 3.839 .965 -4.461 .981 .123 675 1.0 1.7000 3.538 3.862 .858 -4.081 .984 .107 -611 1.0 1.7500 3.635 3.882 .759 -3.862 .986 .092 -549 1.0 1.7750 3.630 3.915 .576 -3.278 .990 .068 -4.33 1.0 1.8000 3.928 3.928 4.499 -2.957 .992 .057 -379 1.0 1.8500 4.126 3.950 .366 -2.305 .994 .041 -286 1.0 1.8650 4.124 3.958 .310 -2.100 .995 .037 -267 1.0 1.8675 4.274 3.962 .285 -1.926 .995 .031 -2267 1.0 <td></td>	
1.6500 3.346 3.814 1.079 -4.693 .978 .140 739 1.00 1.6750 3.442 3.839 .965 -4.461 .981 .123 675 1.0 1.7000 3.538 3.862 .858 -4.081 .984 .107 611 1.0 1.7750 3.635 3.882 .759 -3.862 .986 .092 549 1.0 1.7750 3.830 3.915 .576 -3.278 .990 .068 433 1.0 1.8000 3.928 3.928 .499 -2.957 .992 .057 379 1.0 1.8250 4.126 3.950 .366 -2.305 .994 .041 -2.86 1.0 1.8625 4.175 3.950 .366 -2.305 .994 .041 -2.86 1.0 1.8625 4.126 3.958 .310 -2.100 .995 .037 -267 1.0 1.8625 4.126 3.958 .310 -2.100 .995 .034 -249 1.0	
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1.7000 3.538 3.862 .858 -4.081 .984 .107 611 1.0 1.7250 3.635 3.882 .759 -3.862 .986 .092 549 1.0 1.7500 3.635 3.990 .664 -3.697 .988 .079 489 1.0 1.8000 3.928 3.991 .576 -3.278 .990 .068 433 1.0 1.8250 4.027 3.940 .429 -2.957 .992 .057 372 1.0 1.8550 4.126 3.950 .366 -2.305 .994 .041 286 1.0 1.8750 4.224 3.954 .338 -2.275 .995 .037 267 1.0 1.8875 4.274 3.962 .285 -1.926 .995 .031 230 1.0 1.9125 4.373 3.965 .262 -1.811 .996 .029 213 1.0 1.9250 4.423 3.971 .220 -1.519 .997 .024 179 1.0	
1.7500 3.733 3.900 .664 -3.697 .988 .079 489 1.079 1.7750 3.830 3.915 .576 -3.278 .990 .068 433 1.0 1.8000 3.928 .499 -2.957 .992 .057 379 1.0 1.8250 4.027 3.940 .429 -2.680 .993 .049 -332 1.0 1.86500 4.126 3.950 .366 -2.305 .994 .041 286 1.0 1.8650 4.224 3.958 .310 -2.100 .995 .037 267 1.0 1.8650 4.224 3.958 .310 -2.100 .995 .034 249 1.0 1.8650 4.224 3.958 .310 -2.100 .995 .034 249 1.0 1.8675 4.224 3.958 .310 -2.100 .995 .031 230 1.0 1.9000 4.324 3.965 .2262 -1.811 .996 .029 -213 1.0 1.9250 4.423 3.971 .220 -1.519 .997 .024 165 1.0 1.9500 4.522	13 .966
1.7750 3.830 3.915 .576 -3.278 .990 .068 433 1.0 1.8000 3.928 3.928 .499 -2.957 .992 .057 -3.79 1.0 1.8250 4.126 3.950 .366 -2.305 .994 .041 -2.86 1.0 1.8550 4.126 3.954 .338 -2.275 .995 .037 267 1.0 1.8750 4.224 3.958 .310 -2.100 .995 .037 267 1.0 1.8875 4.274 3.962 .285 -1.926 .995 .031 230 1.0 1.9125 4.573 3.965 .262 -1.811 .996 .029 213 1.0 1.9250 4.423 3.971 .220 -1.519 .997 .024 179 1.0 1.9500 4.522 3.976 .184 -1.320 .997 .021 165 1.0 1.9504 4.522 3.976 .184 -1.210 .997 .018 133 1.0 1.9625 4.572 3.980 .154 -1.143 .997 .016 128 1.0 1.9875	
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1.8250 4.027 3.940 .429 -2.680 .993 .049 332 1.0 1.8500 4.126 3.950 .366 -2.305 .994 .041 286 1.0 1.8625 4.175 3.958 .338 -2.275 .995 .037 267 1.0 1.8675 4.224 3.958 .310 -2.100 .995 .034 249 1.0 1.9000 4.524 3.962 .285 -1.926 .995 .031 230 1.0 1.9000 4.524 3.965 .262 -1.811 .996 .029 -213 1.0 1.9125 4.373 3.968 .240 -1.673 .996 .026 195 1.0 1.9375 4.472 5.973 .201 -1.438 .997 .024 179 1.0 1.9500 4.522 3.978 .168 -1.210 .997 .019 151 1.0 1.9625 4.572 3.980 .154 -1.143 .997 .016 -128 1.0 </td <td></td>	
$\begin{array}{cccccccccccccccccccccccccccccccccccc$	
$\begin{array}{cccccccccccccccccccccccccccccccccccc$	04 .987
1.8875 4.274 3.962 .285 -1.926 .995 .031 230 1.0 1.9000 4.524 3.965 .262 -1.811 .996 .029 213 1.0 1.9125 4.423 3.971 .220 -1.519 .996 .026 195 1.0 1.9525 4.423 3.971 .220 -1.519 .997 .024 179 1.0 1.9550 4.522 3.973 .201 -1.438 .997 .021 165 1.0 1.9625 4.572 3.978 .168 -1.210 .997 .018 139 1.0 1.9875 4.672 3.980 .154 -1.143 .997 .016 128 1.0 1.9875 4.672 3.982 .140 -1.010 .998 .014 116 1.0 2.0000 4.721 3.983 .128 886 .998 .013 106 1.0	04 .988
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1.9250 4.423 3.971 .220 -1.519 .997 .024 179 1.0 1.9375 4.472 3.973 .201 -1.438 .997 .021 165 1.0 1.9500 4.522 3.976 .184 -1.320 .997 .019 151 1.0 1.9625 4.572 3.980 .154 -1.210 .997 .018 139 1.0 1.9875 4.672 3.982 .140 -1.010 .998 .014 116 1.0 2.0000 4.721 3.983 .128 886 .998 .013 106 1.0	
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1.9750 4.622 3.980 .154 -1.143 .997 .016 128 1.0 1.9875 4.672 3.982 .140 -1.010 .998 .014 116 1.0 2.0000 4.721 3.983 .128 886 .998 .013 106 1.0	
1.9875 4.672 3.982 .140 -1.010 .998 .014 116 1.0 2.0000 4.721 3.983 .128 886 .998 .013 106 1.0	
2.0000 4.721 3.983 .128886 .998 .013106 1.0	
2.0125 4 771 3 004 118 022 000 010 000 1	
2.0125 4.771 3.984 .118 833 .998 .012 098 1.0	.996
2.0250 4.821 3.986 .108 744 .998 .011 089 1.0 2.0375 4.871 3.987 .099 667 .998 .010 081 1.0	
2.0375 4.871 3.987 .099 667 .998 .010 081 1.00 2.0500 4.921 3.988 .091 718 .999 .009 073 1.00	
2.0625 4.971 3.989 .082635 .999 .008068 1.0	
2.0750 5.021 3.990 .075 563 .999 .007 061 1.0	998
2.0875 5.070 3.991 .068 506 .999 .006 055 1.00	
2.1000 5.120 3.992 .062 444 .999 .006 050 1.00 2.1125 5.170 3.992 .057 398 .999 .005 045 1.00	
2.1125 5.170 3.992 .057 398 .999 .005 045 1.00 2.1250 5.220 3.993 .052 359 .999 .004 040 1.00	
2.1375 5.270 3.994 .048 315 .999 .004 037 1.00	
2.1500 5.320 3.994 .044 278 .999 .004 033 1.00	
2.1625 5.370 3.994 .041 248 .999 .003 030 1.00	00 999
2.1750 5.420 3.995 .038 223 .999 .003 026 1.00	.999
2.1875 5.470 3.995 .024 200 .999 .002 023 1.00 2.2000 5.520 3.996 .020 174 .999 .002 021 1.00	
2.2125 5.570 3.996 .017150 .999 .002019 1.00	
2.2250 5.620 3.996 .014 132 .999 .002 018 1.00	
2.2500 5.720 3.997 .011 104 .999 .001 014 1.00	.999
2.2750 5.820 3.997 .008 085 .999 .001 010 1.00	.999
2.3000 5.920 3.998 .006 060 .999 .001 008 1.00 2.3250 6.020 3.998 .005 045 .999 .001 006 1.00	
2.3250 6.020 3.998 .005 045 .999 .001 006 1.00 2.3500 6.120 3.998 .004 036 .999 .000 004 1.00	
2.3750 6.220 3.999 .003 032 .999 .000 002 1.00	
2.4000 6.320 3.999 .002 016 .999 .000 002 1.00	0 1.000
2.4250 6.420 4.000 .002 006 .999 .000 002 1.00	0 1.000
2.4500 6.520 4.000 .002 .002 .999 .000002 1.00	0 1.000

TABLE II - SUMMARY OF HEAT-TRANSFER AND FRICTION PARAMETERS AND BOUNDARY-LAYER THICKNESSES

NACA

						744	
f _W	T ₈₈	Eu	Nu √Re θ¹w	C _f /Re f _w "	δ*√Re x	$\delta_1 \frac{\sqrt{Re}}{x}$	δ _c √Re/x
0	1	-0.0904 0868 0826 0741 0654 0476 .00 .50	0.1982 .2214 .2310 .2435 .2528 .2673 .2926 .4162 .4958	0 .0580 .0870 .1296 .1637 .2202 .3320 .8997 1.2326	3.498 2.972 2.762 2.510 2.336 2.092 1.721 .855 .648	0.868 .853 .838 .812 .788 .746 .662 .374	0.626 .693 .720 .752 .773 .801 .834 .792
	2	-0.1178 09 05 .00 .50	0.1890 .2522 .2756 .2944 .4002 .4726	0 .1634 .2434 .3125 .6794 .8987	4.582 2.430 1.882 1.537 .699 .515	1.664 1.501 1.383 1.271 .899 .763	1.076 1.408 1.478 1.495 1.370 1.215
	4	-0.1351 09 05 .00 .50 1.00	0.1794 .2642 .2790 .2952 .3876 .4530	0 .1934 .2397 .2874 .5367 .6854	6.950 2.297 1.810 1.428 .588 .427	3.109 2.719 2.582 2.457 1.887 1.615	1.834 2.595 2.651 2.663 2.344 2.075
- 1	1	-0.0418 .00 .50	0.1029 .1662 .2594 .2934	0 .1648 .6974 .9692	4.272 2.459 1.033 .783	0.954 .827 .444 .345	0.807 .973 .994 .918
	2	.50 1.00	0.1602 .2290 .2526	0.1476 .4733 .6344	2.381 .877 .637	1.605 1.117 .968	1.778 1.760 1.613
	4	-0.0644 .00 .50 1.00	.0796 .1506 .2006 .2134	0 .1263 .3309 .4222	7.219 2.460 .773 .553	3.484 3.100 2.486 2.235	2.620 3.236 3.123 2.861
-1	1	-0.0072 .00 .05 .15 .50	0.0251 .0516 .0880 .1128 .1392 .1456	0 .0355 .1410 .2703 .5344 .7565	6.398 4.396 2.796 2.008 1.252	1.116 1.073 .911 .750 .524 .405	1.072 1.147 1.241 1.280 1.269 1.208
	2	0 .05 .15 .40 .50	0.0406 .0692 .0886 .1028 .1044 .1024	0.0242 .0892 .1678 .2866 .3205 .4408	4.931 2.985 1.989 1.245 1.114 .793	2.109 1.908 1.710 1.476 1.428 1.289	2.167 2.299 2.343 2.320 2.289 2.206
	4	0 .05 .15 .40 .50	0.0262 .0510 .0656 .0726 .0718	0.0125 .0542 .1030 .1705 .1882 .2463	6.409 3.405 2.040 1.156 1.016	4.161 3.859 3.640 3.362 3.299 3.186	4.002 4.248 4.309 4.266 4.254 4.101
0	1/2	-0.06 -0.04 .00 .50	0.2064 .2554 .2900 .4412 .5298	0 .1735 .3462 1.2754 1.8000	3.043 2.309 1.898 .980 .768	0.441 .406 .347 .116 .065	0.348 .420 .457 .460 .415
	1	0 1 2 1	0.2884 .4801 .5812	0.3556 1.9299 2.7842	2.031 1.033 .820	0.182 030 064	0.246 .271 .246

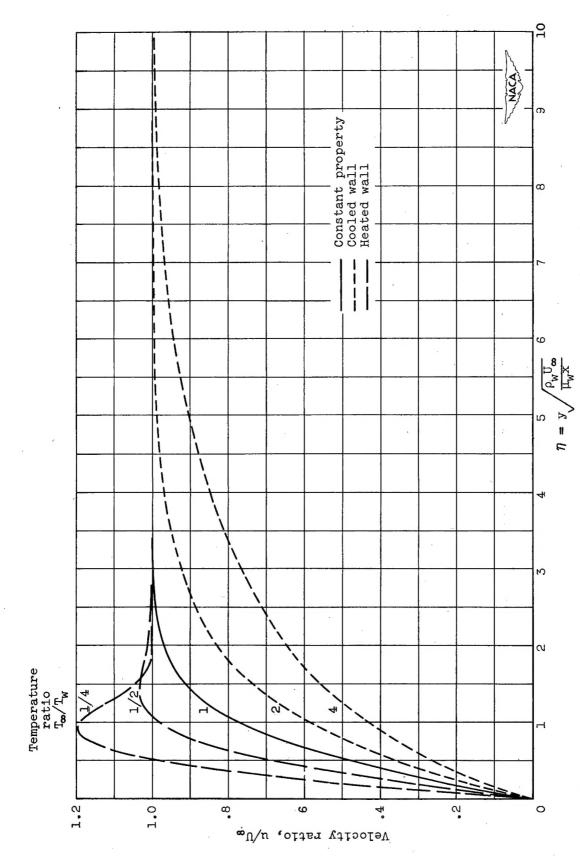


Figure 1. - Velocity distribution in boundary layer for impermeable wall and Euler number of 1.

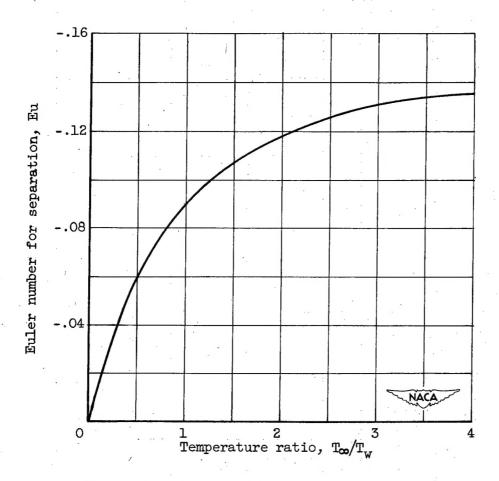


Figure 2. - Values of Euler number for separation from impermeable wall.

1. Flow, Laminar (1.1.3.1) 2. Heat Transfer, Aerodynamic (1.1.4.2) 3. Turbine Cooling (3.7.2) 4. Heat Transfer Theory and Experiment (3.9.1) I. Brown, W. Byron II. Donoughe, Patrick L. III. NACA TN 2479	1. Flow, Laminar (1.1.3.1) 2. Heat Transfer, Aerodynamic (1.1.4.2) 3. Turbine Cooling (3.7.2) 4. Heat Transfer Theory and Experiment (3.9.1) I. Brown, W. Byron II. Donoughe, Patrick L. III. NACA TN 2479	NACA
NACA TN 2479 National Advisory Committee for Aeronautics. TABLES OF EXACT LAMINAR-BOUNDARY-LAYER SOLUTIONS WHEN THE WALL IS POROUS AND FLUID PROPERTIES ARE VARIABLE. W. Byron Brown and Patrick L. Donoughe. September 1951. 68p. diagrs., 2 tabs. (NACA TN 2479) Exact solutions of the laminar boundary equations were computed and tabulated for a range of fixed values of Euler number, temperature ratio, and flow through a porous wall. Euler numbers are 0, 0.5, 1, and negative values to the separation point. Temperature ratios are 1, 2, and 4 for the impermeable wall and for two values of coolant flow. In addition, results from temperature ratios of 1/2 and 1/4 are given for the impermeable wall. For each case, boundary-layer thicknesses and heat-transfer and friction coefficients were computed and tabulated.	NACA TN 2479 National Advisory Committee for Aeronautics. TABLES OF EXACT LAMINAR-BOUNDARY-LAYER SOLUTIONS WHEN THE WALL IS POROUS AND FLUID PROPERTIES ARE VARIABLE. W. Byron Brown and Patrick L. Donoughe. September 1951. 68p. diagrs., 2 tabs. (NACA TN 2479) Exact solutions of the laminar boundary equations were computed and tabulated for a range of fixed values of Euler number, temperature ratio, and flow through a porous wall. Euler numbers are 0, 0.5, 1, and negative values to the separation point. Temperature ratios are 1, 2, and 4 for the impermeable wall and for two values of coolant flow. In addition, results from temperature ratios of 1/2 and 1/4 are given for the impermeable wall. For each case, boundary-layer thicknesses and heat-transfer and friction coefficients were computed and tabulated.	Copies obtainable from NACA, Washington
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